parts. The actions would be required to be accomplished in accordance with the SB described previously and in accordance with certain sections of the engine manuals.

Economic Analysis

There are approximately 2,310 engines of the affected design in the worldwide fleet. The FAA estimates that (1) 1,183 engines installed on airplanes of U.S. registry would be affected by this proposed AD, (2) it would take approximately 1 work hour per engine to accomplish the proposed actions, and (3) the average labor rate is $60 per work hour. Required parts would cost approximately $1,465 per engine. Review of purchase order documents indicate that approximately 1,547 pressure tubes have been sold to the airlines; therefore this action would affect only 763 engines. Based on these figures, the total cost impact of the proposed AD on U.S. operators is estimated to be $1,163,575.

Regulatory Impact

This proposal does not have federalism implications, as defined in Executive Order No. 13132, because it would not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Accordingly, the FAA has not consulted with state authorities prior to publication of this proposal.

For the reasons discussed above, I certify that this proposed regulation (1) is not a ''significant regulatory action'' under Executive Order 12866; (2) is not a ''significant rule'' under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES.**

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

Pratt & Whitney: Docket No. 99–NE–25–AD.

Applicability: Pratt & Whitney (PW) JT9D–3A, –7, –7A, –7AH, –7H, –7F, –7J, –7Q, –7Q3, –20, –20J, –59A, –70A, and –7R4D series turbofan engines, installed on but not limited to Boeing 747 and 767 and McDonnell Douglas DC–10 series aircraft.

Note 1: This airworthiness directive (AD) applies to each engine identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For engines that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent oil fires in and around the No. 4 bearing area, which could result in excessive growth due to heat of the sixth stage low pressure turbine (LPT) disk, liberation of the sixth stage LPT disk, uncontained engine failure, and damage to the airplane, accomplish the following:

Installation of Improved Hardware

(a) At the next time when the ''N'' or ''P'' flange is disconnected after the effective date of this AD, install an improved No. 4 bearing internal oil pressure tube in accordance with PW Service Bulletin (SB) No. 5707, dated September 17, 1986, and SB JT9D–7R4–72–289, dated March 26, 1986.

Inspections

(b) Perform initial and repetitive inspections of the No. 4 bearing oil pressure tube and turbine exhaust case (TEC) strut for clearance and alignment, and, if necessary, replace with serviceable parts, in accordance with the applicable PW JT9D Engine Manuals, part numbers (P/Ns) 646028, 777210, and 754459, Turbine Exhaust Case Inspection 01, Section 72–53–01, and P/N 785059, Turbine Exhaust Case Inspection 01, Section 72–53–05, as follows:

(1) Initially inspect at the next time when the ''N'' or ''P'' flange is disconnected after the effective date of this AD.

(2) Thereafter, inspect at each time when the ''N'' or ''P'' flange is disconnected.

Alternative Methods of Compliance

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Engine Certification Office. Operators shall submit their request through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Engine Certification Office.

Note 2: Information concerning the existence of approved alternative methods of compliance with this airworthiness directive, if any, may be obtained from the Engine Certification Office.

Ferry Flights

(d) Special flight permits may be issued in accordance with sections §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Issued in Burlington, Massachusetts, on October 18, 1999.

David A. Downey,

Assistant Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. 99–30630 Filed 11–23–99; 8:45 am]

BILLING CODE 4910–13–U

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 99–NM–132–AD]

RIN 2120–AA64

Airworthiness Directives; Boeing Model 767 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain Boeing Model 767 series airplanes. This proposal would require repetitive inspections of the side load underwing fitting bushings for broken sealant or bushing migration, and corrective action, if necessary. This proposal would also provide optional terminating action in lieu of repetitive inspections. This proposal is prompted by reports of migrated bushings and corrosion on the side load fittings. The actions specified by the proposed AD are intended to prevent corrosion in the side load underwing fitting, which could result in cracking and consequent reduced structural integrity of the wing strut.

DATES: Comments must be received by January 10, 2000.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM–114, Attention: Rules Docket No. 99–NM–132–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056. Comments may be inspected at this location between 9:00 a.m. and 3:00 p.m., Monday through Friday, except Federal holidays.

The service information referenced in the proposed rule may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124–2207. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington.

FOR FURTHER INFORMATION CONTACT: James G. Rehrl, Aerospace Engineer, Airframe Branch, ANM–120S, FAA, Transport Airplane Directorate, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055–4056; telephone (425) 227–2783; fax (425) 227–1181.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this notice may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must submit a self-addressed, stamped postcard on which the following statement is made: ''Comments to Docket Number 99–NM–132–AD.'' The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM–114, Attention: Rules Docket No. 99–NM–132–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056.

Discussion

The FAA has received reports of migrated bushings on the side load underwing fittings on Boeing Model 767 series airplanes. Where migrated bushings were found, several cases of corrosion of the side fitting lug bore were reported. Migration of the side load fitting bushing breaks the moisture barrier to the fitting bore, and can cause corrosion of the fitting bore. Bushing migration can also cause damage to the adjacent structure when the bushing contacts the adjacent structure as it migrates outward. This condition, if not corrected, could result in cracking and consequent reduced structural integrity of the wing strut.

Explanation of Relevant Service Information

The FAA has reviewed and approved Boeing Service Bulletin 767–57–0063, dated May 7, 1998, which describes procedures for repetitive detailed visual inspections of the side load underwing fitting bushings for broken sealant or bushing migration, and corrective action, if necessary. The corrective action includes reworking the side load fitting and installing a new bushing, if necessary. The service bulletin also describes optional terminating action to rework the side load fitting that may be accomplished in lieu of repetitive detailed visual inspections.

Explanation of Requirements of Proposed Rule

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same type design registered in the United States, the proposed AD would require accomplishment of the actions specified in the service bulletin described previously. This proposed AD also would provide for optional terminating action for the repetitive inspections.

Operators should note that the FAA has determined that the repetitive inspections proposed by this AD can be allowed to continue in lieu of accomplishment of a terminating action. In making this determination, the FAA considers that, in this case, long-term continued operational safety will be adequately assured by accomplishing the repetitive inspections to detect broken sealant or bushing migration of the side load fitting bushing before it represents a hazard to the airplane.

Differences Between the Proposed Rule and the Service Bulletin

Boeing Service Bulletin 767–57–0063, dated May 7, 1998, specifies that the manufacturer may be contacted for disposition of certain repair conditions, this proposed AD would require the repair of those conditions to be accomplished in accordance with a method approved by the FAA, or in accordance with data meeting the type certification basis of the airplane approved by a Boeing Company Designated Engineering Representative who has been authorized by the FAA to make such findings. For a repair method to be approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA, Transport Airplane Directorate, as specified in paragraph (d) of this proposed AD, the Manager's approval letter must specifically reference this AD.

Cost Impact

There are approximately 663 airplanes of the affected design in the worldwide fleet. The FAA estimates that 312 airplanes of U.S. registry would be affected by this proposed AD, that it would take approximately 4 work hours per airplane to accomplish the proposed inspection, and that the average labor rate is $60 per work hour. Based on these figures, the cost impact of the proposed AD on U.S. operators is estimated to be $74,880, or $240 per airplane, per inspection cycle.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

Should an operator elect to accomplish the optional terminating action rather than continue the repetitive inspections, it would take approximately 12 work hours per airplane to accomplish the modification, at an average labor rate of $60 per work hour.

Required parts would cost approximately $1,500 per airplane. Based on these figures, the cost impact of this optional terminating action is estimated to be $2,220 per airplane.

Regulatory Impact

The regulations proposed herein would not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order

12612, it is determined that this proposal would not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this proposed regulation (1) is not a ''significant regulatory action'' under Executive Order 12866; (2) is not a ''significant rule'' under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption ''ADDRESSES.''

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

Boeing: Docket 99–NM–132–AD.

Applicability: Model 767 series airplanes, as listed in Boeing Service Bulletin 767–57–0063, dated May 7, 1998; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (f) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent corrosion in the side load underwing fitting, which could result in cracking and consequent reduced structural integrity of the wing strut, accomplish the following:

Initial Inspection

(a) Accomplish a detailed visual inspection of the side load underwing fitting to detect broken sealant or bushing migration, in accordance with Boeing Service Bulletin 767–57–0063, dated May 7, 1998, at the later of the times specified in paragraphs (a)(1) and (a)(2) of this AD.

(1) Within 10 years since the date of manufacture, or

(2) Within 3000 cycles or 18 months after the effective date of this AD, whichever occurs first.

Note 2: For the purposes of this AD, a detailed visual inspection is defined as: ''An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required.''

Repetitive Inspections

(b) If no broken sealant or evidence of bushing migration is detected, repeat the inspection required by paragraph (a) of this AD thereafter at intervals not to exceed 6 years.

Corrective Action

(c) If evidence of broken sealant or bushing migration is detected, prior to further flight, accomplish the corrective action specified in paragraph (c)(1) or (c)(2) of this AD, except as provided by paragraph (d) of this AD.

(1) Remove the bushing and inspect the side load fitting lug bore for corrosion as shown in Figure 3 of Boeing Service Bulletin 767–57–0063, dated May 7, 1998, and rework as specified in Figure 3 of the service bulletin.

(2) Reseal the bushing and service the side load fitting in accordance with Boeing Service Bulletin 767–57–0063, dated May 7, 1998. Within 3,000 flight cycles or 18 months, whichever occurs first, remove the bushing and inspect the side load fitting lug bore for corrosion as shown in Figure 3 of the service bulletin, and rework as specified in Figure 3 of the service bulletin.

(d) For airplanes on which broken sealant or evidence of bushing migration is detected: During the accomplishment of the actions specified in paragraphs(c)(1) or (c)(2) of this AD, if damage to the lug bore or fitting is found that is outside the limits specified in Boeing Service Bulletin 767–57–0063, dated May 7, 1998; prior to further flight, repair in accordance with a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA, Transport Airplane Directorate; or in accordance with data meeting the type certification basis of the airplane approved by a Boeing Company Designated Engineering Representative who has been authorized by the FAA to make such findings. For a repair method to be approved by the Manager, Seattle ACO, as required by this AD, the Manager's approval letter must specifically reference this AD.

Optional Terminating Action

(e) Accomplishment of rework of the side load fitting in accordance with Figure 1 or Figure 3 of Boeing Service Bulletin 767–57–0063, dated May 7, 1998, constitutes terminating action for the requirements of this AD.

Alternative Methods of Compliance

(f) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Seattle ACO.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Seattle ACO.

Special Flight Permits

(g) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Issued in Renton, Washington, on November 18, 1999.

D.L. Riggin,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 99–30631 Filed 11–23–99; 8:45 am]

BILLING CODE 4910–13–U

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 99–NM–309–AD]

RIN 2120–AA64

Airworthiness Directives; British Aerospace (Jetstream) Model 4101 Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to all British Aerospace (Jetstream) Model 4101 airplanes. This proposal would require manufacture and installation of a placard on the left-hand instrument panel in the cockpit to prohibit push-backs of the airplane while the engines are running. In lieu of accomplishing