

(3) If any electrical arcing or exposed copper conductor is detected, prior to further flight, accomplish either paragraph (b)(3)(i) or (b)(3)(ii) of this AD.

(i) Replace any section of the electrical conduit where the arcing occurred with a new section, in accordance with the alert service bulletin, and accomplish the requirements specified in paragraph (b)(1)(ii) of this AD.

(ii) Perform a visual inspection to detect fuel leaks of the electrical conduit, in accordance with the alert service bulletin.

(A) If no fuel leak is detected, prior to further flight, accomplish the requirements specified in paragraph (b)(1)(ii) of this AD. Repeat the inspection required by paragraph (b)(3)(ii) of this AD thereafter at intervals not to exceed 1,500 flight hours, until the replacement required by paragraph (b)(3)(ii)(B) of this AD is accomplished.

(B) If any fuel leak is detected, prior to further flight, replace any section of the electrical conduit where the leak is with a new section, in accordance with the alert service bulletin. Prior to further flight after accomplishment of the replacement, accomplish the requirements specified in paragraph (b)(1)(ii) of this AD. Accomplishment of electrical conduit replacement constitutes terminating action for the repetitive inspection requirements of paragraph (b)(3)(ii)(A) of this AD.

(4) If any presence or scent of fuel on the electrical wires is detected, prior to further flight, locate the source of the leak and replace the damaged conduit with a new conduit, in accordance with the alert service bulletin; and accomplish the requirements specified in either paragraph (b)(1)(i) or (b)(1)(ii) of this AD, unless accomplished previously in accordance with paragraph (b)(1), (b)(2), or (b)(3) of this AD.

(c) Accomplish the requirements specified in either paragraph (c)(1) or (c)(2) of this AD, in accordance with Part 2 of the Accomplishment Instructions of Boeing Alert Service Bulletin 737-28A1132, dated December 2, 1998, or Revision 1 dated January 15, 1999.

(1) Deactivate the center tank float switch (i.e., cut the two wires for the float switch at the splices on the front spar and cap and stow the four wire ends); paint a Caution that shows a conservative maximum fuel capacity for the center tank on the underside of the right-hand wing near the fueling station door; and install an INOP placard on the fueling panel.

(2) Deactivate the center tank float switch (i.e., cut, stow, and splice the two wires for the float switch at the splices on the front spar), and paint a Caution that shows a conservative maximum fuel capacity for the center tank on the underside of the right-hand wing near the fueling station door.

(d) For airplanes on which the requirements specified in paragraph (c) of this AD have been accomplished: Accomplish the requirements specified in paragraph (d)(1), (d)(2), and (d)(3) of this AD.

(1) Operators must ensure that airplane fueling crews are properly trained in accordance with the procedures specified in Boeing Telex M-7200-98-04486, dated December 1, 1998, or procedures approved

by the FAA. This one-time training must be accomplished prior to utilizing the procedures specified in paragraph (d)(3) of this AD.

(2) Prior to fueling the airplane, perform a check to verify that the fueling panel center tank quantity indicator is operative. Repeat this check thereafter prior to fueling the airplane. If the fueling panel center tank quantity indicator is not operative, prior to further flight, replace the fueling panel center tank quantity indicator with a serviceable part.

(3) One of the two manual fueling procedures for the center fuel tank must be used for each fueling occurrence, in accordance with Boeing Telex M-7200-98-04486, dated December 1, 1998, or a method approved by the FAA.

**Note 2:** For the purposes of this AD, the term "the FAA," is defined in paragraph (d) of this AD as "the cognizant Principal Maintenance Inspector (PMI)."

**Note 3:** Where there are differences between the Boeing Alert Service Bulletin 737-28A1132 and this AD, the AD prevails.

(e) Dispatch with the center fuel tank float switch deactivated, in accordance with Boeing Alert Service Bulletin 737-28A1132, dated December 2, 1998, or Revision 1, dated January 15, 1999, is allowed until replacement float switches and wiring are available for installation. Where there are differences between the Master Minimum Equipment List (M MEL) and the AD, the AD prevails.

(f) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA PMI, who may add comments and then send it to the Manager, Seattle ACO.

**Note 4:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Seattle ACO.

(g) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(h) Except as provided by paragraphs (d)(1) and (d)(2) of this AD, the actions shall be done in accordance with Boeing Alert Service Bulletin 737-28A1132, dated December 2, 1998; Boeing Alert Service Bulletin 737-28A1132, Revision 1, dated January 15, 1999; and Boeing Telex M-7200-98-04486, dated December 1, 1998, as applicable. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124-2207. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(i) This amendment becomes effective on March 18, 1999.

Issued in Renton, Washington, on February 23, 1999.

**Darrell M. Pederson,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 99-5042 Filed 3-2-99; 8:45 am]

BILLING CODE 4910-13-U

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

RIN 2120-AA64

[Docket No. 97-SW-14-AD; Amendment 39-11062; AD 99-05-14]

#### Airworthiness Directives; Eurocopter France Model SA. 315B, SA. 316B, SA. 316C, SA. 319B, and SE. 3160 Helicopters

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to Eurocopter France Model SA. 315B, SA. 316B, SA. 316C, SA. 319B, and SE. 3160 helicopters, that requires inspecting the main rotor blade cuff attachment fitting in the area of the main rotor blade (blade) attachment bolts for cracks, and removing and replacing the blade if a crack is found. This amendment is prompted by a report of a crack in a blade cuff attachment fitting/spar assembly that was discovered during fatigue testing by the manufacturer. The actions specified by this AD are intended to prevent failure of a blade cuff attachment fitting at a bolt hole location, loss of a blade, and subsequent loss of control of the helicopter.

**EFFECTIVE DATE:** April 7, 1999.

**FOR FURTHER INFORMATION CONTACT:** Richard Monschke, Aerospace Engineer, FAA, Rotorcraft Directorate, Rotorcraft Standards Staff, 2601 Meacham Blvd., Fort Worth, Texas 76137, telephone (817) 222-5116, fax (817) 222-5961.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to Eurocopter France Model SA. 315B, SA. 316B, SA. 316C, SA. 319B, and SE. 3160 helicopters was published in the **Federal Register** on November 3, 1998 (63 FR 59252). That action proposed to require inspecting the blade cuff attachment fitting in the area of the blade attachment bolt holes

for cracks, and removing and replacing any blade in which a crack is found.

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were received on the proposal or the FAA's determination of the cost to the public. The FAA has determined that air safety and the public interest require the adoption of the rule with only minor editorial changes that will neither increase the economic burden on any operator nor increase the scope of the AD.

The FAA estimates that 83 helicopters of U.S. registry will be affected by this AD, that it will take approximately 2 work hours per helicopter for the initial inspection and 2 work hours per helicopter for each repetitive inspection and that the average labor rate is \$60 per work hour. Required parts will cost approximately \$40,000 per blade, if needed. Based on these figures, the total cost impact of the AD on U.S. operators is estimated to be \$49,960 for one inspection and one blade replacement for each helicopter per year.

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this final rule does not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a

substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the FAA, Office of the Regional Counsel, Southwest Region, Attention: Rules Docket No. 97-SW-14-AD, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

#### Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

2. Section 39.13 is amended by adding a new airworthiness directive to read as follows:

#### AD 99-05-14 Eurocopter France:

Amendment 39-11062. Docket No. 97-SW-14-AD.

**Applicability:** Model SA. 315B, SA. 316B, SA. 316C, SA. 319B, and SE. 3160 helicopters, with a main rotor blade, part number (P/N) 3160S.11.10.000, 3160S.11.30.000, 3160S.11.35.000, 3160S.11.40.000, 3160S.11.45.000, 3160S.11.50.000, or 3160S.11.55.000, installed, certificated in any category.

**Note 1:** This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area

subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must use the authority provided in paragraph (c) to request approval from the FAA. This approval may address either no action, if the current configuration eliminates the unsafe condition, or different actions necessary to address the unsafe condition described in this AD. Such a request should include an assessment of the effect of the changed configuration on the unsafe condition addressed by this AD. In no case does the presence of any modification, alteration, or repair remove any helicopter from the applicability of this AD.

**Compliance:** (1) For blades with less than 400 hours time-in-service (TIS), required prior to the accumulation of 400 hours TIS, unless accomplished previously, and thereafter at intervals not to exceed 400 hours TIS; or (2) for blades with 400 hours or more TIS, required within 50 hours TIS or 30 calendar days after the effective date of this AD, whichever occurs first, unless accomplished previously, and thereafter at intervals not to exceed 400 hours TIS:

To prevent failure of a main rotor blade (blade) cuff attachment fitting at a bolt hole location, loss of a blade, and subsequent loss of control of the helicopter, accomplish the following:

(a) Inspect both upper and lower blade surfaces of each blade cuff for cracks (see Figure 1) as follows:

(1) Use a mild liquid detergent or equivalent to remove all dirt from the blade cuff.

(2) Inspect the blade cuff for cracks, paying particular attention to the area around the attaching bolts, using a 10-power or higher magnifying glass.

(3) If a crack is suspected, remove any paint and clean the area under inspection using a Naptha-type solvent or equivalent, and conduct a dye penetrant inspection. Completely isolate the area under inspection with self-adhesive aluminum tape to prevent solvent or penetrating dye seepage into the other areas of the blade.

(b) If a crack is detected, remove the blade and replace it with an airworthy blade.

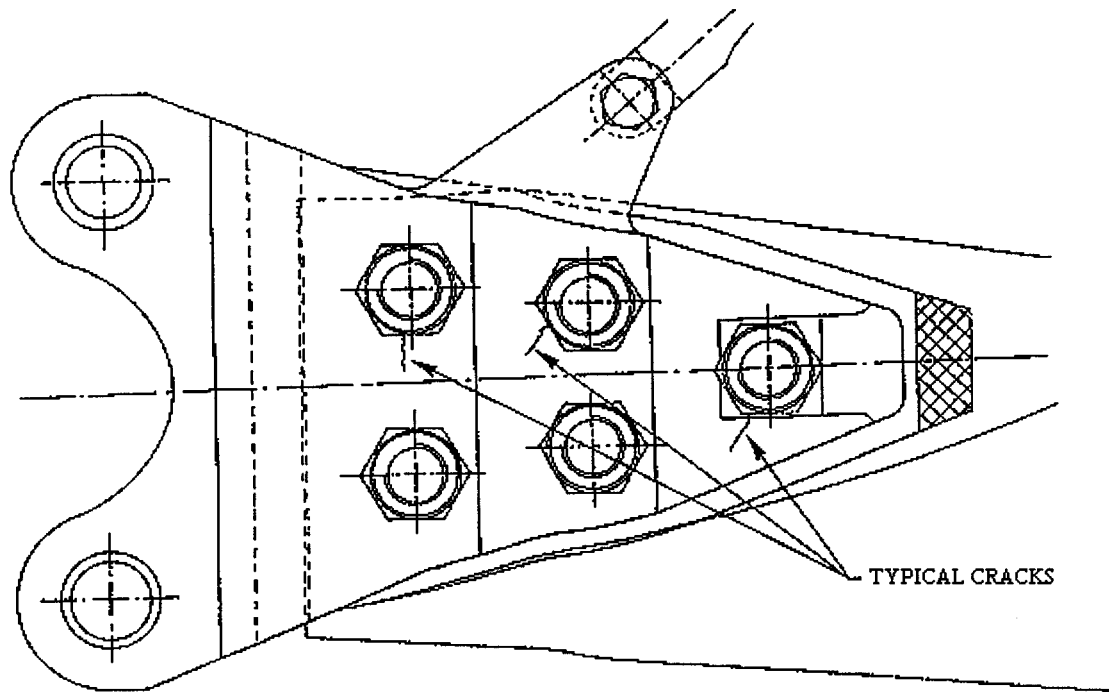


Figure 1

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Rotorcraft Standards Staff, Rotorcraft Directorate, FAA. Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment and then send it to the Manager, Rotorcraft Standards Staff.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Rotorcraft Standards Staff.

(d) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the helicopter to a location where the requirements of this AD can be accomplished.

(e) This amendment becomes effective on April 7, 1999.

**Note 3:** The subject of this AD is addressed in Direction Generale De L'Aviation Civile (France) AD 96-081-036(B)R1, and AD 96-082-054(B)R1, both dated April 24, 1996.

Issued in Fort Worth, Texas, on February 24, 1999.

**Henry A. Armstrong,**

*Manager, Rotorcraft Directorate, Aircraft Certification Service.*

[FR Doc. 99-5178 Filed 3-2-99; 8:45 am]

BILLING CODE 4910-13-U

## DEPARTMENT OF THE TREASURY

### Internal Revenue Service

#### 26 CFR Parts 1 and 602

[TD 8795]

RIN 1545-AT78

#### Notice of Significant Reduction in the Rate of Future Benefit Accrual; Correction

**AGENCY:** Internal Revenue Service (IRS), Treasury.

**ACTION:** Correction to final regulations.

**SUMMARY:** This document contains corrections to Treasury Decision 8795, which was published in the **Federal Register** on Monday, December 14, 1998 (63 FR 68678) relating to defined benefit plans and to individual account plans that are subject to the funding standards of section 302 of the Employment Retirement Income Security Act of 1974.

**DATES:** These corrections are effective December 14, 1998.

#### FOR FURTHER INFORMATION CONTACT:

Diane S. Bloom, (202) 622-6214 or Christine L. Keller, (202) 622-6090 (not toll-free numbers).

#### SUPPLEMENTARY INFORMATION:

#### Background

The final regulations that are the subject of these corrections are under

section 411 of the Internal Revenue Code.

#### Need for Correction

As published, TD 8795 contains errors which may prove to be misleading and are in need of clarification.

#### Correction of Publication

Accordingly, the publication of the final regulations (TD 8795), which was the subject of FR Doc. 98-32925, is corrected as follows:

1. On page 68680, column 2, in the preamble under the paragraph heading "Special Analyses", line 12, the language "24, 1996, the Regulatory Flexibility Act" is corrected to read "29, 1996, the Regulatory Flexibility Act".

#### § 602.101 [Corrected]

2. On page 68684, column 1, § 602.101(c), in the table under the column heading Current OMB control No., the OMB number "1545-1447" is corrected to read "1545-1477".

**Cynthia E. Grigsby,**

*Chief, Regulations Unit, Assistant Chief Counsel (Corporate).*

[FR Doc. 99-5129 Filed 3-2-99; 8:45 am]

BILLING CODE 4830-01-U