

that no operator would accomplish those actions in the future if this AD were not adopted.

Should an operator elect to accomplish the modification, it would take approximately 5 work hours per airplane to accomplish the actions, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the optional modification provided by this AD on U.S. operators is estimated to be \$300 per airplane.

### Regulatory Impact

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this final rule does not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption ADDRESSES.

### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

### Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

### § 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

**98-18-24 Airbus Industrie:** Amendment 39-10740. Docket 97-NM-156-AD.

**Applicability:** Model A320 series airplanes on which Airbus Modification 21778 (reference Airbus Service Bulletin A320-53-1072, dated November 7, 1995, as revised by Change Notice 0A, dated July 5, 1996) has not been accomplished, certificated in any category.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To correct fatigue cracking in the inner flange of door frame 66, left and right, which could result in reduced structural integrity of the airplane, accomplish the following:

(a) Prior to the accumulation of 20,000 total flight cycles, or within 1 year after the effective date of this AD, whichever occurs later: Perform a rotating probe eddy current inspection to detect cracking around the edges of the gusset plate attachment holes of the inner flange of door frame 66, left and right, at stringer positions P18, P20, P22, P18, P20, and P22, in accordance with Airbus Service Bulletin A320-53-1071, dated November 7, 1995, as revised by Change Notice 0A, dated July 5, 1996. If any crack is detected, prior to further flight, repair in accordance with a method approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate. Repeat the inspection thereafter at intervals not to exceed 20,000 flight cycles.

(b) Modification of the gusset plate attachment holes of the inner flange of door frame 66, left and right (Airbus Modification 21778), in accordance with Airbus Service Bulletin A320-53-1072, dated November 7, 1995, as revised by Change Notice 0A, dated July 5, 1996, constitutes terminating action for the repetitive inspection requirements of this AD.

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

**Note 2:** Information concerning the existence of approved alternative methods of

compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

(d) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(e) The inspections shall be done in accordance with Airbus Service Bulletin A320-53-1071, dated November 7, 1995, as revised by Change Notice 0A, dated July 5, 1996. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**Note 3:** The subject of this AD is addressed in French airworthiness directive 96-234-087(B), dated October 20, 1996.

(f) This amendment becomes effective on October 20, 1998.

Issued in Renton, Washington, on August 28, 1998.

**Vi L. Lipski,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*  
[FR Doc. 98-24248 Filed 9-14-98; 8:45 am]  
BILLING CODE 4910-13-U

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 97-NM-290-AD; Amdt. 39-10741; AD 98-18-25]

RIN 2120-AA64

### irworthiness Directives; Fokker Model F28 Mark 1000, 2000, 3000, and 4000 Series Airplanes

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to certain Fokker Model F28 Mark 1000, 2000, 3000, and 4000 series airplanes, that requires replacement of certain hinges on the forward, center, and aft cargo doors with improved hinges. This amendment is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by this AD are intended to prevent failure of the cargo door hinges caused by stress corrosion or fatigue cracks, which could result in

decompression of the airplane, and possible in-flight separation of the cargo door.

**DATES:** Effective October 20, 1998.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of October 20, 1998.

**ADDRESSES:** The service information referenced in this AD may be obtained from Fokker Services B.V., Technical Support Department, P.O. Box 75047, 1117 ZN Schiphol Airport, the Netherlands. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Norman B. Martenson, Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington, 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Fokker Model F28 Mark 1000, 2000, 3000, and 4000 series airplanes was published in the **Federal Register** on December 18, 1997 (62 FR 66317). That action proposed to require replacement of certain hinges on the forward, center, and aft cargo doors with improved hinges.

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the single comment received.

#### **Request To Require Revision 12 of Structural Integrity Program (SIP)**

One commenter suggests that the FAA revise AD 91-05-10 to require accomplishment of Revision 12 of the F28 Structural Integrity Program (SIP), rather than Revision 10. The commenter states that this change would be more effective than issuance of the proposed AD, which requires replacement of the cargo door hinges in accordance with Fokker Service Bulletin F28/52-110, dated April 7, 1993. The commenter notes that, as part of SIP Items 52-30-09 and 52-30-10, Revision 12 of the SIP specifies a reduction in the inspection intervals for the cargo door hinges, following their replacement as described in Fokker Service Bulletin F28/52-110. The commenter states that

this reduction indicates that the hinges installed per the service bulletin are not significantly improved over those previously installed, and that the actions required by this proposed AD may be obsolete.

The FAA does not concur with the commenter's request to revise AD 91-05-10 and withdraw this proposed AD. The FAA first finds it necessary to clarify that AD 93-13-04, amendment 39-8617 (58 FR 38513, July 19, 1993), presently requires accomplishment of Revision 10 of the SIP, rather than AD 91-05-10, as suggested by the commenter. Based on information provided by the manufacturer, as well as further review of SIP Items 52-30-09 and 52-30-10, the FAA has determined that replacement of the cargo door hinges is necessary, as required by this AD, in order to adequately address the identified unsafe condition. The FAA may also consider separate rulemaking to require accomplishment of Revision 12 of the SIP; however, no change to this final rule is necessary.

#### **Conclusion**

After careful review of the available data, including the comment noted above, the FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

#### **Cost Impact**

The FAA estimates that 37 airplanes of U.S. registry will be affected by this AD.

It will take approximately 62 work hours per airplane to replace the forward cargo door hinge, at an average labor rate of \$60 per work hour. Required parts will cost approximately \$5,740 per airplane. Based on these figures, the cost impact of this replacement required by this AD on U.S. operators is estimated to be \$350,020, or \$9,460 per airplane.

It will take approximately 62 work hours per airplane to replace the center cargo door hinge, at an average labor rate of \$60 per work hour. Required parts will cost approximately \$5,650 per airplane. Based on these figures, the cost impact of this replacement required by this AD on U.S. operators is estimated to be \$346,690, or \$9,370 per airplane.

It will take approximately 46 work hours per airplane to replace the aft cargo door hinge, at an average labor rate of \$60 per work hour. Required parts will cost approximately \$6,470 per airplane. Based on these figures, the cost impact of this replacement required by this AD on U.S. operators is estimated to be \$341,510, or \$9,230 per airplane.

The cost impact figures discussed above are based on assumptions that no

operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

#### **Regulatory Impact**

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this final rule does not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

#### **List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### **Adoption of the Amendment**

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### **PART 39—AIRWORTHINESS DIRECTIVES**

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### **§ 39.13 [Amended]**

2. Section 39.13 is amended by adding the following new airworthiness directive:

**98-18-25 Fokker:** Amendment 39-10741. Docket 97-NM-290-AD.

*Applicability:* Model F28 Mark 1000, 2000, 3000, and 4000 series airplanes; serial numbers 11003 through 11241 inclusive, 11991, and 11992; certificated in any category.

**Note 1:** This AD applies to each airplane identified in the preceding applicability

provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent failure of the cargo door hinges caused by stress corrosion and/or fatigue cracks, which could result in decompression of the airplane, and possible in-flight separation of the cargo door; accomplish the following:

(a) Within 12 months after the effective date of this AD, replace the hinges on the forward, center, and aft belly cargo doors with improved hinges in accordance with Part 1, Part 2, and Part 3, as applicable, of the Accomplishment Instructions of Fokker Service Bulletin F28/52-110, dated April 7, 1993.

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

(c) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(d) The actions shall be done in accordance with Fokker Service Bulletin F28/52-110, dated April 7, 1993. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Fokker Services B.V., Technical Support Department, P.O. Box 75047, 1117 ZN Schiphol Airport, the Netherlands. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**Note 3:** The subject of this AD is addressed in Dutch airworthiness directive 93-055 (A), dated April 23, 1993.

(e) This amendment becomes effective on October 20, 1998.

Issued in Renton, Washington, on August 28, 1998.

**Vi L. Lipski,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 98-24249 Filed 9-14-98; 8:45 am]

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 96-NM-123-AD; Amendment 39-10737; AD 98-18-21]

RIN 2120-AA64

#### Airworthiness Directives; Construcciones Aeronauticas, S.A. (CASA) Model C-212 Series Airplanes

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to all CASA Model C-212 series airplanes, that requires implementation of a corrosion prevention and control program either by accomplishing specific inspections or by revising the maintenance inspection program to include such a program. This amendment is prompted by reports of incidents involving corrosion and fatigue cracking in transport category airplanes that are approaching or have exceeded their economic design goal; these incidents have jeopardized the airworthiness of the affected airplanes. The actions specified by this AD are intended to prevent degradation of the structural capabilities of the airplane due to the problems associated with corrosion.

**DATES:** Effective October 20, 1998.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of October 20, 1998.

**ADDRESSES:** The service information referenced in this AD may be obtained from Construcciones Aeronauticas, S.A., Getafe, Madrid, Spain. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Norman B. Martenson, Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601

Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to all CASA Model C-212 series airplanes was published in the **Federal Register** on February 5, 1997 (62 FR 5350). That action proposed to require implementation of a corrosion prevention and control program either by accomplishing specific inspections or by revising the maintenance inspection program to include such a program.

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comments received.

#### Request To Shorten Initial Compliance Time

Several commenters request that the one year compliance time for accomplishment of initial corrosion inspections, as specified in the proposed AD, be shortened to be effective immediately upon issuance of the AD. The commenters consider the one year period for implementation of the corrosion prevention and control program (CPCP) to be too long, unnecessary, and not in the best interests of public safety.

The FAA does not concur with the commenters' request. In developing an appropriate compliance time, the FAA considered the risk to the affected airplanes, as well as the magnitude and complexity of the CPCP. The FAA does not consider the risk to these airplanes during the one year implementation period to be great, since the requirement to implement the CPCP does not stem from a specific finding of serious corrosion on CASA Model C-212 series airplanes. Rather, the CPCP is proactive in nature, in that it establishes a comprehensive program designed to prevent corrosion from developing in the future to the point that it could affect safe operation of these airplanes.

However, the FAA does consider it necessary to allow operators sufficient time for implementation of the requirements of the CPCP. The tasks to be accomplished as part of the CPCP are complex and time consuming; complete accomplishment of these tasks could require an elapsed time of several weeks. Given the magnitude of the CPCP tasks required by this AD, the FAA considers a one year period to be appropriate, to allow operators time to plan for implementation of these tasks on the fleet of affected airplanes.