

this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To detect and correct corrosion or plating cracks of the pin assemblies in the front trunnion support of the main landing gear (MLG), which could cause these assemblies to break and result in collapse of the MLG, accomplish the following:

(a) Perform a close visual inspection to detect corrosion or plating cracks of each 4330M Steel pin assembly in the forward trunnion support of the MLG, in accordance with Boeing Alert Service Bulletin 767-57A0047, Revision 1, dated May 9, 1996, at the later of the times specified in paragraphs (a)(1) and (a)(2) of this AD.

(1) Within 4 years since date of manufacture of the airplane, or 4 years since the last overhaul of the MLG. Or

(2) Within 18 months after the effective date of this AD.

(b) If no corrosion or crack is detected, repeat the close visual inspection thereafter at intervals not to exceed 48 months.

(c) If any corrosion or crack is detected, prior to further flight, replace it with a new pin assembly made from 15-5PH CRES with Class 3 chrome plating, in accordance with Boeing Alert Service Bulletin 767-57A0047, Revision 1, dated May 9, 1996.

(d) Accomplishment of replacement of a 4330M Steel pin assembly with a new pin assembly made from 15-5PH CRES with Class 3 chrome plating, in accordance with Boeing Alert Service Bulletin 767-57A0047, Revision 1, dated May 9, 1996, constitutes terminating action for the inspections required by this AD for that pin location.

Note 2: Replacement of a 4330M Steel pin assembly with a new pin assembly made from 15-5PH CRES with Class 3 chrome plating prior to the effective date of this AD, in accordance with Boeing Service Bulletin 767-57A0047, dated January 19, 1995, is considered an acceptable method of compliance with paragraph (d) of this AD for that pin location.

(e) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Seattle ACO.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Seattle ACO.

(f) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(g) The inspections and replacement shall be done in accordance with Boeing Alert Service Bulletin 767-57A0047, Revision 1, dated May 9, 1996. This incorporation by reference was approved by the Director of the

Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124-2207. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(h) This amendment becomes effective on September 4, 1998.

Issued in Renton, Washington, on July 24, 1998.

S.R. Miller,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 98-20340 Filed 7-30-98; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 98-NM-116-AD; Amendment 39-10687; AD 98-16-11]

RIN 2120-AA64

Airworthiness Directives; Airbus Model A300, A310, and A300-600 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Airbus Model A300, A310, and A300-600 series airplanes, that requires repetitive detailed visual inspections to detect cracks in the pylon thrust and sideload fitting of the wing, and replacement of any cracked pylon thrust and sideload fitting with a new fitting. This amendment is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by this AD are intended to detect and correct cracks in the pylon thrust and sideload fitting of the wing, which could result in reduced structural integrity of the airplane.

DATES: Effective September 4, 1998.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of September 4, 1998.

ADDRESSES: The service information referenced in this AD may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation

Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Norman B. Martenson, Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Airbus Model A300, A310, and A300-600 series airplanes was published in the **Federal Register** on May 28, 1998 (63 FR 29153). That action proposed to require repetitive detailed visual inspections to detect cracks in the pylon thrust and sideload fitting of the wing, and replacement of any cracked pylon thrust and sideload fitting with a new fitting.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the single comment received.

The commenter supports the proposed rule.

Conclusion

After careful review of the available data, including the comment noted above, the FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Cost Impact

The FAA estimates that 126 airplanes of U.S. registry will be affected by this AD, that it will take approximately 3 work hours per airplane to accomplish the required inspection, and that the average labor rate is \$60 per work hour. Based on these figures, the cost impact of the AD on U.S. operators is estimated to be \$22,680, or \$180 per airplane, per inspection cycle.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

Regulatory Impact

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or

on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this final rule does not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption ADDRESSES.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

98-16-11 Airbus Industrie: Amendment 39-10687. Docket 98-NM-116-AD.

Applicability: Model A300 series airplanes, as listed in Airbus Service Bulletin A300-57-0232, Revision 01, dated January 12, 1998; Model A310 series airplanes, as listed in Airbus Service Bulletin A310-57-2075, Revision 01, dated January 12, 1998; and Model A300-600 series airplanes, as listed in Airbus Service Bulletin A300-57-6079, Revision 02, dated January 12, 1998; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an

alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To detect and correct cracks in the pylon thrust and sideload fitting of the wing, which could result in reduced structural integrity of the airplane, accomplish the following:

(a) Prior to the accumulation of 2,800 total flight cycles, or within 18 months after the effective date of this AD, whichever occurs later, perform a detailed visual inspection to detect cracks in the pylon thrust and sideload fitting of the wing, in accordance with Airbus Service Bulletin A300-57-0232, Revision 01 (for Model A300 series airplanes); A310-57-2075, Revision 01 (for Model A310 series airplanes); or A300-57-6079, Revision 02 (for Model A300-600 series airplanes); all dated January 12, 1998; as applicable. Repeat the detailed visual inspection thereafter at intervals not to exceed 2,800 flight cycles.

(b) If any crack is detected during any inspection required by paragraph (a), prior to further flight, replace the pylon thrust and sideload fitting with a new fitting in accordance with Airbus Service Bulletin A300-57-0232, Revision 01 (for Model A300 series airplanes); A310-57-2075, Revision 01 (for Model A310 series airplanes); or A300-57-6079, Revision 02 (for Model A300-600 series airplanes); all dated January 12, 1998; as applicable. Thereafter, continue the inspections in accordance with the requirements of paragraph (a) of this AD.

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

(d) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(e) The inspections and replacement shall be done in accordance with Airbus Service Bulletin A300-57-0232, Revision 01, dated January 12, 1998; Airbus Service Bulletin A310-57-2075, Revision 01, dated January 12, 1998; or Airbus Service Bulletin A300-57-6079, Revision 02, dated January 12, 1998; as applicable. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex,

France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 3: The subject of this AD is addressed in French airworthiness directive 97-358-232(B), dated November 19, 1997.

(f) This amendment becomes effective on September 4, 1998.

Issued in Renton, Washington, on July 24, 1998.

S.R. Miller,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 98-20434 Filed 7-30-98; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 98-NM-90-AD; Amendment 39-10686; AD 98-16-10]

RIN 2120-AA64

Airworthiness Directives; Dornier Model 328-100 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Dornier Model 328-100 series airplanes, that requires revising the Airplane Flight Manual (AFM) to require use of the electrical fuel pump for take-off and landing and performance of an operational check of the electrical fuel pump following landing. This amendment also requires replacement of the jet booster pumps with new or modified jet booster pumps, which terminates the requirement for the AFM revision. This amendment is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by this AD are intended to prevent failure of the jet booster pumps, which could result in reduced engine thrust during take-off or landing, and consequent increased risk of impact with terrain.

DATES: Effective September 4, 1998.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of September 4, 1998.

ADDRESSES: The service information referenced in this AD may be obtained from FAIRCHILD DORNIER, DORNIER