

implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption ADDRESSES.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

2. Section 39.13 is amended by adding a new airworthiness directive to read as follows:

#### AD 98-12-15 Eurocopter France:

Amendment 39-10571. Docket No. 98-SW-07-AD.

**Applicability:** Model AS 332C, L, L1, and L2 helicopters, with intermediate gearboxes (IGB), part numbers (P/N) 332A35-0002 all dash numbers, 332A35-0010 all dash numbers, or 332A35-0011-01, installed, except those IGBs modified in accordance with MOD 0761049 or MOD 0761050, certificated in any category.

**Note 1:** This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must use the authority provided in paragraph (c) to request approval from the FAA. This approval may address either no action, if the current configuration eliminates the unsafe condition, or different actions necessary to address the unsafe condition described in this AD. Such a request should include an assessment of the

effect of the changed configuration on the unsafe condition addressed by this AD. In no case does the presence of any modification, alteration, or repair remove any helicopter from the applicability of this AD.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent failure of the IGB-to-structure attachment stirrup (stirrup) front tabs, loss of anti-torque drive, and subsequent loss of control of the helicopter, accomplish the following:

(a) Before the first flight of each day, perform a visual inspection of the stirrup front tabs for cracks in accordance with paragraph 2.B.1) of the Accomplishment Instructions in Eurocopter France AS 332 Service Bulletin 01.00.47, Revision No. 1, dated September 10, 1997 (SB). If a crack is found, remove the IGB and replace it with an airworthy IGB before further flight. Completion of the conformity procedure contained in paragraph 2.B.2.1.3) of the SB is terminating action for the requirement of this AD to inspect for cracks prior to the first flight of each day.

(b) Within 100 hours time-in-service (TIS), inspect the two front attaching assemblies securing the stirrup of the IGB to the angle bracket of the structure (attachment assembly) for thickness of the stirrup front tabs in accordance with paragraph 2.B.2) of the SB.

(1) If the attachment assembly meets the conformity requirements of either paragraph 2.B.2.1.1) or 2.B.2.1.2) of the SB, reassemble the attachment assembly in accordance with paragraph 2.B.2.1.3) of the SB.

(2) If the attachment assembly does not meet the conformity requirements of either paragraph 2.B.2.1.1) or 2.B.2.1.2) of the SB, replace it with an attachment assembly which does meet the conformity requirements of either of those paragraphs. Install the attachment assembly hardware in accordance with 2.B.2.1.3) of the SB.

(3) If a crack is discovered in the stirrup front tabs as a result of the conformity inspection, remove the IGB and replace it with an airworthy IGB before further flight.

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, FAA, Rotorcraft Directorate, Rotorcraft Standards Staff. Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment and then send it to the Manager, Rotorcraft Standards Staff.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Rotorcraft Standards Staff.

(d) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the helicopter to a location where the requirements of this AD can be accomplished.

(e) The inspections and replacement, if necessary, shall be done in accordance with Eurocopter France AS 332 Service Bulletin 01.00.47, Revision No. 1, dated September 10, 1997. This incorporation by reference was approved by the Director of the Federal

Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from American Eurocopter Corporation, 2701 Forum Drive, Grand Prairie, Texas 75053-4005. Copies may be inspected at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(f) This amendment becomes effective on July 14, 1998.

**Note 3:** The subject of this AD is addressed in Direction Generale De L'Aviation Civile (France) AD 96-263-060(AB)R1 for Eurocopter France (ECF) Model AS 332C, L, and L1 helicopters, and AD 96-262-004(AB)R1 for ECF Model AS 332L2 helicopters, both dated November 5, 1997.

Issued in Fort Worth, Texas, on May 29, 1998.

**Henry A. Armstrong,**

*Manager, Rotorcraft Directorate, Aircraft Certification Service.*

[FR Doc. 98-15124 Filed 6-8-98; 8:45 am]

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 97-SW-07-AD; Amendment 39-10572; AD 98-12-16]

RIN 2120-AA64

#### Airworthiness Directives; Eurocopter France Model SA 330F, G, and J Helicopters

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to Eurocopter France Model SA 330F, G, and J helicopters that requires visually inspecting the intermediate gearbox (IGB) fairing safety stop (safety stop) for cracks, crazing, or edge wear, and if a crack, crazing, or edge wear exceeds the established limits, replacing the safety stop; and, inspecting to ensure that the inclined drive shaft fairing hinge pin is properly locked. A terminating action is provided in the AD by installing an additional safety stop on the IGB fairing. This amendment is prompted by one report of an accident involving the loss of the inclined drive shaft fairing. The actions specified by this AD are intended to prevent loss of the inclined drive shaft fairing, impact with the tail rotor, and subsequent loss of control of the helicopter.

**DATES:** Effective July 14, 1998.

The incorporation by reference of certain publications listed in the

regulations is approved by the Director of the Federal Register as of July 14, 1998.

**ADDRESSES:** The service information referenced in this AD may be obtained from American Eurocopter Corporation, 2701 Forum Drive, Grand Prairie, Texas 75053-4005. This information may be examined at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC. **FOR FURTHER INFORMATION CONTACT:** Mr. Mike Mathias, Aerospace Engineer, FAA, Rotorcraft Directorate, Rotorcraft Standards Staff, 2601 Meacham Blvd., Fort Worth, Texas 76137, telephone (817) 222-5123, fax (817) 222-5961.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to Eurocopter France Model SA 330F, G, and J helicopters was published in the **Federal Register** on January 22, 1998 (63 FR 3273). That action proposed to require visually inspecting the IGB safety stop for cracks, crazing, or edge wear, and if a crack, crazing, or edge wear exceeds the established limits, replacing the safety stop; and, inspecting to ensure that the inclined drive shaft fairing hinge pin is properly locked. A terminating action was provided in the AD by installing an additional safety stop on the IGB fairing.

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were received on the proposal or the FAA's determination of the cost to the public. The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

The FAA estimates that 1 helicopter of U.S. registry will be affected by this AD, that it will take approximately 1 work hour to perform the inspection and two work hours to install the safety stop, and that the average labor rate is \$60 per work hour. Required parts will cost approximately \$50 per helicopter. Based on these figures, the total cost impact of the AD on U.S. operators is estimated to be \$230.

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this final rule does not have sufficient federalism

implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

2. Section 39.13 is amended by adding a new airworthiness directive to read as follows:

##### **AD 98-12-16 Eurocopter France:**

Amendment 39-10572. Docket No. 97-SW-07-AD.

**Applicability:** Model SA 330 F, G, and J helicopters, certificated in any category.

**Note 1:** This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must use the authority provided in paragraph (d) to request approval from the FAA. This approval may address either no action, if the current configuration eliminates the unsafe condition, or different actions necessary to address the unsafe condition described in this AD. Such a request should include an assessment of the effect of the changed configuration on the unsafe condition addressed by this AD. In no case does the presence of any modification, alteration, or repair remove any helicopter from the applicability of this AD.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent loss of the inclined drive shaft fairing hinge pin (hinge pin), that could result in loss of the inclined drive shaft fairing, impact with the tail rotor, and subsequent loss of control of the helicopter, accomplish the following:

(a) Within 7 calendar days after the effective date of this AD, and thereafter, upon the completion of the last flight of each day, visually inspect the intermediate gearbox (IGB) fairing safety stop (safety stop) and the hinge pin in accordance with the Accomplishment Instructions of Eurocopter France SA 330 Service Bulletin No. 54.20, Revision 1, dated February 27, 1996.

(1) Inspect the IGB fairing safety stop, part number (P/N) 330A24-2086-20, for cracks, crazing, and edge wear that exceeds the limits stated in Note II of the Accomplishment Instructions of Eurocopter France SA 330 Service Bulletin No. 54.20, Revision 1, dated February 27, 1996, and if cracks, crazing, or edge wear that exceeds the established limits is detected, remove the safety stop and replace it with an airworthy safety stop; and,

(2) Inspect the hinge pin to ensure it is properly locked.

(b) Within 60 calendar days after the effective date of this AD, install an additional safety stop, P/N 330A24-2119-21, to prevent the hinge pin from backing out of its hole in case of a locking arm failure, in accordance with Accomplishment Instructions of Eurocopter France SA 330 Service Bulletin No. 54.20, Revision 1, dated February 27, 1996.

(c) Installation of an airworthy additional safety stop, P/N 330A24-2119-21, constitutes terminating action for the requirements of this AD.

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Rotorcraft Standards Staff, Rotorcraft Directorate, FAA. Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment and then send it to the Manager, Rotorcraft Standards Staff.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Rotorcraft Standards Staff.

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the helicopter to a location where the requirements of this AD can be accomplished.

(f) The inspection shall be done in accordance with the Accomplishment Instructions of Eurocopter France SA 330 Service Bulletin No. 54.20, Revision 1, dated February 27, 1996. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from American Eurocopter Corporation, 2701 Forum Drive, Grand Prairie, Texas 75053-4005. Copies may be inspected at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the

Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(g) This amendment becomes effective on July 14, 1998.

**Note 3:** The subject of this AD is addressed in Direction Generale De L'Aviation Civile (France) AD 96-095-076(B), dated April 24, 1996.

Issued in Fort Worth, Texas, on May 29, 1998.

**Henry A. Armstrong,**

*Manager, Rotorcraft Directorate, Aircraft Certification Service.*

[FR Doc. 98-15199 Filed 6-8-98; 8:45 am]

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 98-SW-03-AD; Amendment 39-10574; AD 98-12-20]

RIN 2120-AA64

#### **Airworthiness Directives; Eurocopter France Model SE3130, SA3180, SE313B, SA318B, and SA318C Helicopters**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to Eurocopter France Model SE3130, SA3180, SE313B, SA318B, and SA318C helicopters, that requires an initial and repetitive visual inspections and modification, if necessary, of the horizontal stabilizer spar tube (spar tube). This amendment is prompted by an in-service report of fatigue cracks that initiated from corrosion pits. The actions specified by this AD are intended to prevent fatigue failure of the spar tube, separation and impact of the horizontal stabilizer with the main or tail rotor, and subsequent loss of control of the helicopter.

**DATES:** Effective July 14, 1998.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of July 14, 1998.

**ADDRESSES:** The service information referenced in this AD may be obtained from American Eurocopter Corporation, 2701 Forum Drive, Grand Prairie, Texas 75053-4005. This information may be examined at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the

Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Mr. Richard Monschke, Aerospace Engineer, FAA, Rotorcraft Directorate, Rotorcraft Standards Staff, 2601 Meacham Blvd., Fort Worth, Texas 76137, telephone (817) 222-5116, fax (817) 222-5961.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to Eurocopter France Model SE3130, SA3180, SE313B, SA318B, and SA318C helicopters was published in the **Federal Register** on April 21, 1998 (63 FR 19668). That action proposed to require an initial and repetitive visual inspections and modification, if necessary, of the horizontal stabilizer spar tube (spar tube).

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were received on the proposal or the FAA's determination of the cost to the public. The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

The FAA estimates that 14 helicopters of U.S. registry would be affected by this proposed AD, that it would take approximately 0.5 work hour per helicopter to accomplish the inspection and 3 work hours per helicopter to accomplish the modification, and that the average labor rate is \$60 per work hour. Required parts would cost approximately \$1100 per helicopter. Based on these figures, the total cost impact of the proposed AD on U.S. operators is estimated to be \$1310 per helicopter.

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this final rule does not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory

Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

#### **List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety. Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### **PART 39—AIRWORTHINESS DIRECTIVES**

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### **§ 39.13 [Amended]**

2. Section 39.13 is amended by adding a new airworthiness directive to read as follows:

#### **AD 98-12-20 Eurocopter France:**

Amendment 39-10574. Docket No. 98-SW-03-AD.

**Applicability:** SE3130, SA3180, SE313B, SA318B, and SA318C helicopters with horizontal stabilizer, part number (P/N) 3130-35-60-000, 3130-35-60-000-1, 3130-35-60-000-2, 3130-35-60-000-3, 3130-35-60-000-4 or higher dash numbers, installed, certificated in any category.

**Note 1:** This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must use the authority provided in paragraph (f) to request approval from the FAA. This approval may address either no action, if the current configuration eliminates the unsafe condition, or different actions necessary to address the unsafe condition described in this AD. Such a request should include an assessment of the effect of the changed configuration on the unsafe condition addressed by this AD. In no case does the presence of any modification, alteration, or repair remove any helicopter from the applicability of this AD.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent fatigue failure of the horizontal stabilizer spar tube (spar tube), impact of the horizontal stabilizer with the main or tail rotor and subsequent loss of control of the helicopter, accomplish the following:

(a) Before further flight:

(1) Inspect the aircraft records and the horizontal stabilizer installation to determine whether Modification 072214 (installation of the spar tube without play) or Modification