

DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. 97-NM-75-AD; Amendment 39-10353; AD 98-04-42]

RIN 2120-AA64

Airworthiness Directives; Grumman Model TS-2A Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to all Grumman Model TS-2A series airplanes, that requires revising the Airplane Flight Manual (AFM) to modify the limitation that prohibits positioning the power levers below the flight idle stop during flight, and to add a statement of the consequences of such positioning of the power levers. This amendment is prompted by incidents and accidents involving airplanes equipped with turboprop engines in which the ground propeller beta range was used improperly during flight. The actions specified by this AD are intended to prevent loss of airplane controllability, or engine overspeed and consequent loss of engine power, caused by the power levers being positioned below the flight idle stop when the airplane is in flight.

EFFECTIVE DATE: March 30, 1998.

ADDRESSES: Information pertaining to this rulemaking action may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington.

FOR FURTHER INFORMATION CONTACT: Frank Hoerman, Aerospace Engineer, Flight Test Branch, ANM-160L, FAA, Transport Airplane Directorate, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California 90712; telephone (562) 527-5371; fax (562) 625-5210.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to all Grumman Model TS-2A series airplanes was published in the **Federal Register** on December 9, 1997 (62 FR 64780). That action proposed to require revising the Limitations Section of the Airplane Flight Manual (AFM) to modify the limitation that prohibits positioning of the power levers below the flight idle stop while the airplane is in flight, and

adds a statement of the consequences of positioning the power levers below the flight idle stop while the airplane is in flight.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

Conclusion

The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Interim Action

This is considered interim action until final action is identified, at which time the FAA may consider further rulemaking.

Cost Impact

The FAA estimates that 1 airplane of U.S. registry will be affected by this AD, that it will take approximately 1 work hour per airplane to accomplish the required actions, and that the average labor rate is \$60 per work hour. Based on these figures, the cost impact of the AD on the single U.S. operator is estimated to be \$60.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

Regulatory Impact

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12812, it is determined that this final rule does not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy

of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

98-04-42 Grumman: Amendment 39-10353. Docket 97-NM-75-AD.

Applicability: All Model TS-2A series airplanes, certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent loss of airplane controllability, or engine overspeed and consequent loss of engine power, caused by the power levers being positioned below the flight idle stop while the airplane is in flight, accomplish the following:

(a) Within 30 days after the effective date of this AD, revise the Limitations Section of the FAA-approved Airplane Flight Manual (AFM) to include the following statements. This action may be accomplished by inserting a copy of this AD into the AFM.

Positioning of power levers below the flight idle stop while the airplane is in flight is prohibited. Such positioning may lead to loss of airplane control or may result in an overspeed condition and consequent loss of engine power.

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Los Angeles Aircraft Certification Office (ACO),

FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Operations Inspector, who may add comments and then send it to the Manager, Los Angeles ACO.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Los Angeles ACO.

(c) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(d) This amendment becomes effective on March 30, 1998.

Issued in Renton, Washington, on February 12, 1998.

Gilbert L. Thompson,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 98-4248 Filed 2-20-98; 8:45 am]

BILLING CODE 4910-13-U

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 98-NM-30-AD; Amendment 39-10352; AD 98-04-41]

RIN 2120-AA64

Airworthiness Directives; Boeing Model 737-200 and -300 Series Airplanes Equipped With a Main Deck Cargo Door Installed in Accordance With Supplemental Type Certificate SA2969SO

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments.

SUMMARY: This amendment adopts a new airworthiness directive (AD) that is applicable to certain Boeing Model 737-200 and -300 series airplanes. This action requires repetitive inspections to detect cracks in the hinge and lift actuator box area of the main deck cargo door and upper jamb of the fuselage; and repair or replacement of any cracked part with a new part having the same part number. This amendment is prompted by a report that, during a periodic heavy maintenance check, cracks were found in the upper jamb area of the fuselage and in the main deck cargo door. The actions specified in this AD are intended to detect and correct such cracking, which could result in reduced structural integrity of the main cargo door and/or fuselage, and consequent loss or opening of the main deck cargo door while the airplane

is in flight, or reduced controllability of the airplane.

DATES: Effective March 10, 1998.

Comments for inclusion in the Rules Docket must be received on or before April 24, 1998.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 98-NM-30-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

Information concerning this amendment may be obtained from or examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Small Airplane Directorate, Atlanta Aircraft Certification Office, One Crown Center, 1895 Phoenix Boulevard, Suite 450, Atlanta, Georgia 30349.

FOR FURTHER INFORMATION CONTACT:

Curtis Jackson, Aerospace Engineer, Airframe and Propulsion Branch, ACE-117A, FAA, Small Airplane Directorate, Atlanta Aircraft Certification Office, One Crown Center, 1895 Phoenix Boulevard, Suite 450, Atlanta, Georgia 30349; telephone (770) 703-6083; fax (770) 703-6097.

SUPPLEMENTARY INFORMATION: The FAA has received a report that, during a periodic heavy maintenance check of a Boeing Model 737-300 series airplane equipped with a main deck cargo door installed in accordance with Supplemental Type Certificate SA2969SO, cracks were found in the upper jamb area of the fuselage and in the main cargo door. The cracks were between 0.50 inches and 2.35 inches in length. The cause of such cracking is unknown at this time. However, several scenarios (e.g., improper cargo door operations during loading and unloading of cargo, and improper fastener locations) are being examined at this time to determine a possible cause of the cracking.

Cracking in the upper jamb area of the fuselage or in the main deck cargo door, if not corrected, could result in reduced structural integrity of the main deck cargo door and/or fuselage, and consequent loss or opening of the main deck cargo door while the airplane is in flight, or reduced controllability of the airplane.

Explanation of the Requirements of the Rule

Since an unsafe condition has been identified that is likely to exist or develop on other Boeing Model 737-200 and -300 series airplanes, equipped with a main deck cargo door installed in

accordance with Supplemental Type Certificate SA2969SO, of the same type design, this AD is being issued to detect and correct cracking in the upper jamb area of the fuselage and in the main deck cargo door; such cracking could result in reduced structural integrity of the main deck cargo door and/or fuselage, and consequent loss or opening of the main deck cargo door while the airplane is in flight, or reduced controllability of the airplane. This AD requires repetitive detailed visual inspections to detect cracks in the hinge and lift actuator box area of the main deck cargo door and upper jamb of the fuselage; and replacement of any cracked part with a new part having the same part number, or repair in accordance with a method approved by the FAA.

Interim Action

This is considered to be interim action until final action is identified, at which time the FAA may consider further rulemaking.

Determination of Rule's Effective Date

Since a situation exists that requires the immediate adoption of this regulation, it is found that notice and opportunity for prior public comment hereon are impracticable, and that good cause exists for making this amendment effective in less than 30 days.

Comments Invited

Although this action is in the form of a final rule that involves requirements affecting flight safety and, thus, was not preceded by notice and an opportunity for public comment, comments are invited on this rule. Interested persons are invited to comment on this rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified under the caption **ADDRESSES**. All communications received on or before the closing date for comments will be considered, and this rule may be amended in light of the comments received. Factual information that supports the commenter's ideas and suggestions is extremely helpful in evaluating the effectiveness of the AD action and determining whether additional rulemaking action would be needed.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the rule that might suggest a need to modify the rule. All comments submitted will be available, both before and after the closing date for comments,