

elevator trim tab actuators with actuators that have P/N 159SCC100-11, in accordance with Gulfstream Aircraft Service Change No.191, dated August 18, 1972. This installation constitutes terminating action for the inspections required by this AD.

(e)(1) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Atlanta Aircraft Certification Office (ACO), FAA, Small Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Atlanta ACO.

(2) Alternative methods of compliance, approved previously in accordance with AD 72-24-02, amendment 39-1559, are approved as alternative methods of compliance with this AD.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Atlanta ACO.

(f) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Issued in Renton, Washington, on February 27, 1997.

Darrell M. Pederson,

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 97-5460 Filed 3-5-97; 8:45 am]

BILLING CODE 4910-13-U

#### 14 CFR Part 39

[Docket No. 96-CE-25-AD]

RIN 2120-AA64

#### **Airworthiness Directives; Pilatus Britten-Norman Ltd. (formerly Britten-Norman) BN-2A, BN-2B, and BN-2T Series Airplanes**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** This document proposes to adopt a new airworthiness directive that would apply to Pilatus Britten-Norman Ltd. (Pilatus Britten-Norman) BN-2A, BN-2B, and BN-2T series airplanes. The proposed AD would require repetitively inspecting the junction of the torque link lug and upper case of the main landing gear (MLG) torque link assemblies for cracks, and replacing any MLG torque link assembly with a Modification A39 MLG torque link assembly, either immediately when cracks are found or after a certain period of time if cracks are not found. Replacing all MLG torque link assemblies with Modification A39 MLG

torque link assemblies would eliminate the need for the repetitive inspections. These proposed repetitive inspections are currently required by AD 86-07-02 for the BN-2A, BN-2B, and BN-2T series airplanes, as well as the BN2A MK. 111 series airplanes. There are no improved design parts for the BN2A MK. 111 series airplanes. The Federal Aviation Administration (FAA) is issuing in a separate action a proposed revision to AD 86-07-02 to retain the repetitive inspection and replacement (if cracked) requirements for the BN2A MK. 111 series airplanes. The actions specified in the proposed AD are intended to prevent failure of the main landing gear caused by cracks in the torque link area, which could lead to loss of control of the airplane during landing operations.

**DATES:** Comments must be received on or before May 12, 1997.

**ADDRESSES:** Submit comments on the proposal in triplicate to the FAA, Central Region, Office of the Assistant Chief Counsel, Attention: Rules Docket No. 96-CE-25-AD, Room 1558, 601 E. 12th Street, Kansas City, Missouri 64106. Comments may be inspected at this location between 8 a.m. and 4 p.m., Monday through Friday, holidays excepted.

Service information that applies to the proposed AD may be obtained from Pilatus Britten-Norman Limited, Bembridge, Isle of Wight, United Kingdom PO35 5PR; telephone 44-1983 872511; facsimile 44-1983 873246. This information also may be examined at the Rules Docket at the address above. **FOR FURTHER INFORMATION CONTACT:** Mr. Tom Rodriguez, Program Officer, Brussels Aircraft Certification Division, FAA, Europe, Africa, and Middle East Office, c/o American Embassy, B-1000 Brussels, Belgium; telephone (32 2) 508.2717; facsimile (32 2) 230.6899; or Mr. S.M. Nagarajan, Project Officer, Small Airplane Directorate, Airplane Certification Service, FAA, 1201 Walnut, Suite 900, Kansas City, Missouri 64106; telephone (816) 426-6932; facsimile (816) 426-2169.

#### **SUPPLEMENTARY INFORMATION:**

##### Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications should identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking

action on the proposed rule. The proposals contained in this notice may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report that summarizes each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket No. 96-CE-25-AD." The postcard will be date stamped and returned to the commenter.

#### Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Central Region, Office of the Assistant Chief Counsel, Attention: Rules Docket No. 96-CE-25-AD, Room 1558, 601 E. 12th Street, Kansas City, Missouri 64106.

#### Discussion

The FAA has determined that reliance on critical repetitive inspections on aging commuter-class airplanes carries an unnecessary safety risk when a design change exists that could eliminate or, in certain instances, reduce the number of those critical inspections. In determining what inspections are critical, the FAA considers (1) the safety consequences if the known problem is not detected during the inspection; (2) the probability of the problem not being detected during the inspection; (3) whether the inspection area is difficult to access; and (4) the possibility of damage to an adjacent structure as a result of the problem.

These factors have led the FAA to establish an aging commuter-class aircraft policy that requires incorporating a known design change when it could replace a critical repetitive inspection. With this policy in mind, the FAA conducted a review of existing AD's that apply to Pilatus Britten-Norman BN-2A, BN-2B, BN-2T, and BN2A MK. 111 series airplanes. Assisting the FAA in this review were (1) Pilatus Britten-Norman; (2) the Regional Airlines Association (RAA); (3) the Civil Aviation Authority of the

United Kingdom; and (4) several operators of the affected airplanes.

From this review, the FAA has identified AD 86-07-02, Amendment 39-5382, as one which falls under the FAA's aging aircraft policy. AD 86-07-02 currently requires repetitively inspecting the junction of the torque link lug and upper case of the main landing gear (MLG) torque link assemblies for cracks on Pilatus Britten-Norman BN-2A, BN-2B, BN-2T, and BN2A MK. 111 series airplanes, and replacing any cracked part.

Pilatus Britten-Norman has developed a modification that, when incorporated, would eliminate the need for the repetitive inspection requirement of AD 86-07-02 for the Pilatus Britten-Norman BN-2A, BN-2B, and BN-2T series airplanes. The requirements of AD 86-07-02 should still apply for the Pilatus Britten-Norman BN2A MK. 111 series airplanes.

#### Applicable Service Information

Fairey Hydraulics Limited has issued Service Bulletin (SB) 32-4, Issue 4, dated January 30, 1990, which applies to the Pilatus Britten-Norman BN-2A, BN-2B, and BN-2T series airplanes. This SB includes procedures for inspecting the junction of the torque link lug and upper case of the MLG torque link assemblies, and installing new Modification A39 MLG torque link assemblies. Pilatus Britten-Norman SB BN-2/SB.170, Issue 4, dated November 16, 1990, references Fairey Hydraulic Limited SB32-4, Issue 4, dated January 30, 1990.

#### The FAA's Determination

The FAA has examined all available information related to this subject matter and has determined that:

- AD action should be taken for the Pilatus Britten-Norman BN-2A, BN-2B, and BN-2T series airplanes to require the installation of Modification A39 MLG torque link assemblies. The repetitive inspections of the junction of the torque link lug and upper case of the MLG torque link assemblies would still be required until the improved parts are installed; and
- AD 86-07-02 should be revised to remove the BN-2A BN-2B, and BN-2T series airplanes from the applicability of that AD, but retain the actions for the BN2A MK. 111 series airplanes (this is being proposed in a separate action).

#### Explanation of the Provisions of the Proposed AD

Since an unsafe condition has been identified that is likely to exist or develop in other Pilatus Britten-Norman BN-2A, BN-2B, and BN-2T series

airplanes of the same type design, the proposed AD would require repetitively inspecting the junction of the torque link lug and upper case of the MLG torque link assemblies for cracks, and replacing any MLG torque link assembly with a Modification A39 MLG torque link assembly, either immediately when cracks are found or at a certain period of time if cracks are not found. Installation of the improved part would eliminate the need for the repetitive inspections. Accomplishment of the proposed inspections and installation would be in accordance with Fairey Hydraulics Limited SB 32-4, Issue 4, dated January 30, 1990.

#### Cost Impact

The FAA estimates that 112 airplanes in the U.S. registry would be affected by the proposed AD, that it would take approximately 13 workhours per airplane to accomplish the proposed action (1 workhour per inspection and 12 workhours for the installation), and that the average labor rate is approximately \$60 an hour. Parts cost approximately \$6,200 per airplane. Based on these figures, the total cost impact of the proposed AD on U.S. operators is estimated to be \$781,760 or \$6,980 per airplane.

The proposed inspections are currently required on the 112 affected airplanes by AD 86-07-02. The proposed AD would not require any additional inspection requirements over that already required by AD 86-07-02. In addition, the cost figures referenced above are based on the presumption that no affected airplane operator has incorporated the proposed inspection-terminating installation. Pilatus Britten-Norman does not know the number of parts distributed to the affected airplane owners/operators. Numerous sets of parts were sent out to the owners/operators of the affected airplanes, but over the years Pilatus Britten-Norman has not retained these records.

#### The FAA's Aging Commuter Aircraft Policy

The intent of the FAA's aging commuter airplane program is to ensure safe operation of commuter-class airplanes that are in commercial service without adversely impacting private operators. Of the approximately 112 airplanes in the U.S. registry that would be affected by the proposed AD, the FAA has determined that approximately 25 percent are operated in scheduled passenger service by 11 different operators. A significant number of the remaining 75 percent are operated in other forms of air transportation such as air cargo and air taxi.

The proposed action would allow at least 1,000 hours TIS after the effective date of the AD before mandatory accomplishment of the design modification (upon the accumulation of 5,000 hours TIS or within the next 1,000 hours TIS after the effective date of the AD, whichever is later). The average utilization of the fleet for those airplanes in commercial commuter service is approximately 25 to 50 hours TIS per week. Based on these figures, operators of commuter airplanes involved in commercial operation would have to accomplish the proposed modification within 5 to 10 calendar months (at the least) after the proposed AD would become effective. For private owners, who typically operate between 100 to 200 hours TIS per year, this would allow 5 to 10 years (at the least) before the proposed modification would be mandatory. The time it would take those in air cargo/air taxi operations before the proposed action would be mandatory is unknown because of the wide variation between each airplane used in this service. The exact numbers would fall somewhere between the average for commuter operators and private operators.

#### Regulatory Flexibility Determination and Analysis

The Regulatory Flexibility Act of 1980 (RFA) was enacted by Congress to ensure that small entities are not unnecessarily or disproportionately burdened by government regulations. The RFA requires government agencies to determine whether rules would have a "significant economic impact on a substantial number of small entities," and, in cases where they would, conduct a Regulatory Flexibility Analysis in which alternatives to the rule are considered. FAA Order 2100.14A, Regulatory Flexibility Criteria and Guidance, outlines FAA procedures and criteria for complying with the RFA. Small entities are defined as small businesses and small not-for-profit organizations that are independently owned and operated or airports operated by small governmental jurisdictions. A "substantial number" is defined as a number that is not less than 11 and that is more than one-third of the small entities subject to a proposed rule, or any number of small entities judged to be substantial by the rulemaking official. A "significant economic impact" is defined by an annualized net compliance cost, adjusted for inflation, which is greater than a threshold cost level for defined entity types.

The entities that would be affected by this AD are mostly in the portion of Standard Industrial Classification (SIC)

4512, Operators of Aircraft for Hire, classified as "unscheduled." FAA Order 2100.14A sets the size threshold for small entities operating aircraft in this category at nine or fewer aircraft owned and the annualized cost thresholds of at least \$4,975 (1996 dollars) for unscheduled operators. A four-year life for the torque link assembly and capital cost of 15-percent would establish an annualized cost of \$2,445 (1996 dollars). This is less than 50-percent of the threshold cost of \$4,975 per year. In order to incur costs of at least \$4,975, an entity would have to operate three or more of the airplanes referenced in the proposed AD. FAA data shows that only five small entities operate three or more of these airplanes. In addition, this data shows that approximately 60 entities operate the airplanes referenced in the proposed AD, but that only 15 of these entities (one-fourth) operate two or more of these airplanes.

Based on this information, less than one-third of the entities would incur significant operating costs under FAA Order 2100.14A. Therefore, the proposed AD would not significantly affect a number of small entities.

A copy of the full Cost Analysis and Regulatory Flexibility Determination for the proposed action may be examined at the FAA, Central Region, Office of the Assistant Chief Counsel, Attention: Rules Docket No. 96-CE-25-AD, Room 1558, 601 E. 12th Street, Kansas City, Missouri.

#### Regulatory Impact

The regulations proposed herein would not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this proposal would not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action has been placed in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the

location provided under the caption **ADDRESSES**.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

#### The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### **PART 39—AIRWORTHINESS DIRECTIVES**

1. The authority citation for part 39 continues to read as follows:

Authority: 49 USC 106(g), 40113, 44701.

#### **§ 39.13 [Amended]**

2. Section 39.13 is amended by adding a new airworthiness directive (AD) to read as follows:

Pilatus Britten-Norman: Docket No. 96-CE-25-AD.

*Applicability:* Models BN-2, BN-2A, BN-2A-3, BN-2A-6, BN-2A-8, BN-2A-2, BN-2A-9, BN-2A-20, BN-2A-21, BN-2A-26, BN-2A-27, BN-2B-20, BN-2B-21, BN-2B-26, BN-2B-27, and BN-2T airplanes (all serial numbers), certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (f) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

*Compliance:* Required as indicated after the effective date of this AD, unless already accomplished.

To prevent failure of the main landing gear caused by cracks in the torque link assembly area, which could lead to loss of control of the airplane during landing operations, accomplish the following:

(a) Prior to further flight after the effective date of this AD or within the next 100 hours time-in-service (TIS) after the last inspection required by AD 86-07-02, whichever occurs later, and thereafter at intervals not to exceed 100 hours TIS until the installations required by paragraph (c) of this AD are accomplished, inspect the junction of the torque link lug and upper case of all main landing gear (MLG) torque link assemblies for cracks (using a 10-power magnifying glass or by dye penetrant methods). Accomplish these inspections in accordance with the ACCOMPLISHMENT INSTRUCTIONS

section of Fairey Hydraulics Limited Service Bulletin (SB) 32-4, Issue 4, dated January 30, 1990. Pilatus Britten-Norman SB BN-2/SB.170, Issue 4, November 16, 1990, references this service bulletin.

Note 2: These inspections were initially a part of AD 86-07-02, which applied to the BN2A MK. 111 series airplanes as well as the airplanes affected by this AD. The "prior to further flight after the effective date of this AD" compliance time was the original initial compliance time of AD 86-07-02, and is being retained to provide credit and continuity for already-accomplished and future inspections.

(b) If any cracks are found during any of the inspections required by this AD, prior to further flight, replace the MLG torque link assembly with a Modification A39 MLG torque link assembly in accordance with the ACCOMPLISHMENT INSTRUCTIONS section of Fairey Hydraulics Limited SB No. 32-4, Issue 4, dated January 30, 1990.

(1) Repetitive inspections are no longer required when all MLG torque assemblies are replaced with Modification A39 MLG torque link assemblies.

(2) Repetitive inspections may no longer be required on one MLG torque assembly, but still be required on another if all haven't been replaced with a Modification A39 MLG torque link assembly.

(c) Upon the accumulation of 5,000 hours TIS or within the next 1,000 hours TIS after the effective date of this AD, whichever occurs later, unless already accomplished as specified in paragraph (b) of this AD, replace each MLG torque link assembly with a Modification A39 MLG torque link assembly in accordance with of the ACCOMPLISHMENT INSTRUCTIONS section of Fairey Hydraulics Limited SB No. 32-4, Issue 4, dated January 30, 1990.

(d) The intervals between the repetitive inspections required by this AD may be adjusted up to 10 percent of the specified interval to allow accomplishing these actions along with other scheduled maintenance on the airplane.

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the inspection requirements of this AD can be accomplished.

(f) An alternative method of compliance or adjustment of the initial or repetitive compliance times that provides an equivalent level of safety may be approved by the Manager, Brussels Aircraft Certification Division, Europe, Africa, Middle East office, FAA, c/o American Embassy, 1000 Brussels, Belgium. The request should be forwarded through an appropriate FAA Maintenance Inspector, who may add comments and then send it to the Manager, Brussels Aircraft Certification Division.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Brussels Aircraft Certification Division.

(g) All persons affected by this directive may obtain copies of the documents referred to herein upon request to Fairey Hydraulics

Limited, Claverham, Bristol, England; or Pilatus Britten-Norman Limited, Bembridge, Isle of Wight, United Kingdom PO35 5PR, as applicable; or may examine these documents at the FAA, Central Region, Office of the Assistant Chief Counsel, Room 1558, 601 E. 12th Street, Kansas City, Missouri 64106.

Issued in Kansas City, Missouri, on February 24, 1997.

Michael Gallagher,  
Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 97-5471 Filed 3-5-97; 8:45 am]

BILLING CODE 4910-13-U

## 14 CFR Part 39

[Docket No. 96-CE-23-AD]

RIN 2120-AA64

### **Airworthiness Directives; Aviat Aircraft, Inc. Models S-1S, S-1T, S-2, S-2A, S-2S, and S-2B Airplanes (formerly known as Pitts Models S-1S, S-1T, S-2, S-2A, S-2S, and S-2B Airplanes)**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** This document proposes to revise AD 96-12-03, which applies to Aviat Aircraft, Inc. (Aviat) Models S-1S, S-1T, S-2, S-2A, S-2S, and S-2B airplanes that are equipped with aft lower fuselage wing attach fittings incorporating either part number (P/N) 76090, 2-2107-1, or 1-210-102. That AD currently requires repetitively inspecting the aft lower fuselage wing attach fitting on both wings for cracks, and modifying any cracked aft lower fuselage wing attach fitting. Modifying both aft lower fuselage wing attach fittings eliminates the repetitive inspection requirement of AD 96-12-03. Aviat recently started incorporating modified aft lower fuselage wing attach fittings on newly manufactured airplanes. The proposed AD would retain the requirements of AD 96-12-03, but would exempt airplanes that had the modified aft lower fuselage wing attach fittings incorporated at manufacture. The actions specified by the proposed AD are intended to prevent possible in-flight separation of the wing from the airplane caused by a cracked fuselage wing attach fitting.

**DATES:** Comments must be received on or before June 4, 1997.

**ADDRESSES:** Submit comments in triplicate to the Federal Aviation Administration (FAA), Central Region, Office of the Assistant Chief Counsel, Attention: Rules Docket No. 96-CE-23-

AD, Room 1558, 601 E. 12th Street, Kansas City, Missouri 64106. Comments may be inspected at this location between 8 a.m. and 4 p.m., Monday through Friday, holidays excepted.

Service information that applies to the proposed AD may be obtained from Aviat Aircraft, Inc., P.O. Box 1240 (postal service delivery), 672 South Washington Street (express mail), Afton, Wyoming 83110. This information also may be examined at the Rules Docket at the address above.

**FOR FURTHER INFORMATION CONTACT:** Mr. Roger Caldwell, Aerospace Engineer, FAA, Denver Aircraft Certification Office, 26805 E. 68th Avenue, Room 214, Denver, Colorado 80249; telephone (303) 342-1086; facsimile (303) 342-1088.

#### **SUPPLEMENTARY INFORMATION:**

##### **Comments Invited**

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications should identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this notice may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report that summarizes each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket No. 96-CE-23-AD." The postcard will be date stamped and returned to the commenter.

##### **Availability of NPRMs**

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Central Region, Office of the Assistant Chief Counsel, Attention: Rules Docket No. 96-CE-23-AD, Room 1558, 601 E. 12th Street, Kansas City, Missouri 64106.

##### **Events Leading to the Proposed AD**

AD 96-12-03, Amendment 39-9645 (61 FR 28730, June 6, 1996), applies to Aviat Models S-1S, S-1T, S-2, S-2A, S-2S, and S-2B airplanes that are equipped with aft lower fuselage wing attach fittings incorporating either part number (P/N) 76090, 2-2107-1, or 1-210-102. The AD currently requires repetitively inspecting the aft lower fuselage wing attach fitting on both wings for cracks, and modifying any cracked aft lower fuselage wing attach fitting. Modifying both aft lower fuselage wing attach fittings eliminates the repetitive inspection requirement of AD 96-12-03. Accomplishment of the actions required by AD 96-12-03 is in accordance with Aviat Service Bulletin (SB) No. 25, dated April 3, 1996.

Aviat recently started incorporating modified aft lower fuselage wing attach fittings on newly manufactured Models S-1S, S-1T, S-2, S-2A, S-2S, and S-2B airplanes. In addition, Aviat revised SB No. 25 (Revised November 12, 1996) to include this airplane serial number effectivity change.

##### **The FAA's Determination**

After examining the circumstances and reviewing all available information related to the incidents described above, the FAA has determined that (1) those airplanes with modified aft lower fuselage wing attach fittings incorporated at manufacture should be exempt from AD 96-12-03; and (2) AD action should be taken to prevent possible in-flight separation of the wing from the airplane caused by a cracked fuselage wing attach fitting.

##### **Explanation of the Provisions of the Proposed AD**

Since an unsafe condition has been identified that is likely to exist or develop in other Aviat Models S-1S, S-1T, S-2, S-2A, S-2S, and S-2B airplanes of the same type design that are equipped with aft lower fuselage wing attach fittings incorporating either P/N 76090, 2-2107-1, or 1-210-102, the FAA is proposing to revise AD 96-12-03. The proposed AD would retain the requirements of AD 96-12-03, but would exempt airplanes that had the modified aft lower fuselage wing attach fittings incorporated at manufacture. Accomplishment of the proposed AD would be in accordance with Aviat SB No. 25, dated April 3, 1996, Revised November 12, 1996.

##### **Cost Impact**

The FAA estimates that 500 airplanes in the U.S. registry would be affected by the proposed AD, that it would take approximately 2 workhours per airplane