

(iv) For uninstalled disks on or after the effective date of this AD, inspect prior to installation.

(2) Thereafter, perform ECI for cracks at intervals not to exceed 4,000 CIS since last ECI.

(3) Prior to further flight, remove cracked disks and replace with serviceable parts.

(c) Inspect 14th stage HPC disks, P/N's 704314, 789814, and 790214, in accordance with NDIP-858, dated November 7, 1995, attached to PW ASB No. A6232, Revision 1, dated January 11, 1996, as follows:

(1) Perform an initial ECI for cracks as follows:

(i) For disks with 2,000 or more CSN, and 2,000 or more CIS since last shop visit, on the effective date of this AD, inspect within the next 1,000 CIS after the effective date of this AD, or at the next shop visit, whichever occurs first.

(ii) For disks with 2,000 or more CSN, and less than 2,000 CIS since last shop visit, on the effective date of this AD, inspect within 3,000 CIS since the last shop visit, or at the next shop visit, whichever occurs first.

(iii) For disks with 2,000 or more CSN, and no previous shop visits, inspect within 3,000 CIS after the effective date of this AD, or at the next shop visit, whichever occurs first.

(iv) For disks with less than 2,000 CSN on the effective date of this AD, inspect at the next shop visit after the effective date of this AD, but before exceeding 3,000 CSN.

(v) For uninstalled disks on or after the effective date of this AD, inspect prior to installation.

(2) Thereafter, perform ECI for cracks at intervals not to exceed 3,000 CIS since last ECI.

(3) Prior to further flight, remove cracked disks and replace with serviceable parts.

(d) Within 30 days of inspection, report inspection results on the form labeled "14th and 15th Stage HPC Disk Inspection Report," attached to PW NDIP-858, dated November 7, 1995, attached to PW ASB No. A6232, Revision 1, dated January 11, 1996, and PW ASB No. JT9D-7R4-A72-524, dated December 13, 1995, to the office and fax number listed on that form. Reporting requirements have been approved by the Office of Management and Budget and assigned OMB control number 2120-0056.

(e) For the purpose of this AD, a shop visit is defined as separation of the "N" flange.

(f) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Engine Certification Office. The request should be forwarded through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Engine Certification Office.

Note: Information concerning the existence of approved alternative methods of compliance with this airworthiness directive, if any, may be obtained from the Engine Certification Office.

(g) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the aircraft to a location where the requirements of this AD can be accomplished.

Issued in Burlington, Massachusetts, on April 18, 1996.

Jay J. Pardee,

Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. 96-11171 Filed 5-3-96; 8:45 am]

BILLING CODE 4910-13-U

14 CFR Part 39

[Docket No. 96-ANE-02]

Airworthiness Directives; Pratt & Whitney JT8D-200 Series Turbofan Engines

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the superseding of an existing airworthiness directive (AD), applicable to certain Pratt & Whitney JT8D-200 series turbofan engines, that currently requires periodic inspection of fan blades for locked rotors and foreign object damage (FOD), unlocking of shrouds if necessary, lubrication of fan blade shrouds, and dimensional restoration of the fan blade leading edge. This action would add a requirement to install improved design fan blades as terminating action for the inspections. This proposal is prompted by the introduction into service of improved design fan blades. The actions specified by the proposed AD are intended to prevent fan blade failure, which can result in damage to the aircraft.

DATES: Comments must be received by July 5, 1996.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), New England Region, Office of the Assistant Chief Counsel, Attention: Rules Docket No. 96-ANE-02, 12 New England Executive Park, Burlington, MA 01803-5299. Comments may be inspected at this location between 8:00 a.m. and 4:30 p.m., Monday through Friday, except Federal holidays.

The service information referenced in the proposed rule may be obtained from Pratt & Whitney, 400 Main St., East Hartford, CT 06108; telephone (860) 565-6600, fax (860) 565-4503. This information may be examined at the FAA, New England Region, Office of the Assistant Chief Counsel, 12 New England Executive Park, Burlington, MA.

FOR FURTHER INFORMATION CONTACT: Mark A. Rumizen, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New

England Executive Park, Burlington, MA 01803-5299; telephone (617) 238-7137, fax (617) 238-7199.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications should identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this notice may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 96-ANE-02." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, New England Region, Office of the Assistant Chief Counsel, Attention: Rules Docket No. 96-ANE-02, 12 New England Executive Park, Burlington, MA 01803-5299.

Discussion

On June 5, 1995, the Federal Aviation Administration (FAA) issued airworthiness directive (AD) 95-12-19, Amendment 39-9270 (60 FR 31388, June 15, 1995), applicable to certain Pratt & Whitney (PW) JT8D-200 series turbofan engines, to require a periodic inspection of fan blades for locked rotors and foreign object damage (FOD), unlocking of shrouds if necessary, lubrication of the fan blade shrouds, and dimensional restoration of the fan blade leading edge. That action was prompted by the determination that fan blades can fail due to high cycle fatigue (HCF) cracking. This HCF cracking can be

caused by FOD, locked shrouds, which can reduce blade vibratory dampening, and leading edge erosion, which can produce blade flutter. That condition, if not corrected, could result in fan blade failure, which can result in damage to the aircraft.

Since the issuance of that AD, the manufacturer has introduced into service fan blades with an improved design configuration that is more resistant to HCF-induced failures.

The FAA has reviewed and approved the technical contents of PW Alert Service Bulletin (ASB) No. A6241, dated January 25, 1996, that describes procedures for inspection of fan blades for locked rotors and FOD, unlocking of shrouds if necessary, lubrication of fan blade shrouds, and dimensional restoration of the fan blade leading edge. This ASB also provides procedures for modification or replacement of fan blades with an improved design configuration that is more resistant to HCF-induced failures.

Since an unsafe condition has been identified that is likely to exist or develop on other products of this same type design, the proposed AD would supersede AD 95-12-19 to continue to require the inspection and maintenance requirements of that AD, and to add a requirement to modify or install the improved design fan blades as terminating action for those inspections and maintenance requirements.

The FAA estimates that 1,100 engines installed on aircraft of U.S. registry would be affected by this proposed AD, that it would take approximately 19 work hours per engine to accomplish the proposed actions, and that the average labor rate is \$60 per work hour. The FAA also estimates that the parts modification would cost is \$2,720 per engine, which includes a manufacturer's discount of \$1,700 per engine. Based on these figures, the total cost impact of the proposed AD on U.S. operators is estimated to be \$4,246,000.

The regulations proposed herein would not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this proposal would not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44

FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 USC 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by removing amendment 39-9270 (60 FR 31388, June 15, 1995) and by adding a new airworthiness directive to read as follows:

Pratt & Whitney: Docket No. 96-ANE-02.

Supersedes AD 95-12-19, Amendment 39-9270.

Applicability: Pratt & Whitney (PW) Models JT8D-209, -217, -217A, -217C, and -219 turbofan engines that have not incorporated PW Service Bulletin (SB) No. 6193, dated October 31, 1994, or with fan blade, Part Numbers (P/N's) 798821, 798821-001, 808121, 808121-001, 809221, 811821, 851121, 851121-001, 5000021-02, 5000021-022, and 5000021-032 installed. These engines are installed on but not limited to McDonnell Douglas MD-80 aircraft.

Note: This AD applies to each engine identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For engines that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must use the authority provided in paragraphs (d) and (e) to request approval from the FAA. This approval may address either no action, if the current configuration eliminates the unsafe condition, or different actions necessary to address the unsafe condition described in this AD. Such a request should include an assessment of the effect of the changed configuration on the unsafe condition addressed by this AD. In no case does the presence of any modification, alteration, or

repair remove any engine from the applicability of this AD.

Compliance: Required as indicated, unless accomplished previously.

To prevent fan blade failure, which can result in damage to the aircraft, accomplish the following:

(a) Inspect fan blades and shrouds, unlock fan blade shrouds, lubricate fan blade shrouds, restore leading edge dimensions, and modify or install improved design fan blades in accordance with the schedule and procedures described in Parts 1, 2, and 3 of the Accomplishment Instructions of PW Alert Service Bulletin (ASB) No. A6241, dated January 25, 1996.

(b) Modification of fan blades to the improved design configuration or installation of improved design fan blades in accordance with Part 3 of the Accomplishment Instructions of PW ASB No. A6241, dated January 25, 1996, constitutes terminating action to the inspections and maintenance actions described in Parts 1 and 2 of that ASB.

(c) For the purpose of this AD, the accomplishment effective date to be used for determination of compliance intervals, as required by Section 2 of PW ASB No. A6241, dated January 25, 1996, is defined as the effective date of this AD.

(d) For the purpose of this AD, "repair" as specified in Part 3, Paragraph A.(1)(b) of the Accomplishment Instructions of PW ASB No. A6241, dated January 25, 1996, is defined as the refurbishment of fan blades in accordance with Part 3, Paragraph C of the Accomplishment Instructions of PW ASB No. A6241, dated January 25, 1996.

(e) Alternative methods of compliance that have been approved for AD 95-12-19 are applicable for this AD and additional approval is not required.

(f) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Engine Certification Office. The request should be forwarded through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Engine Certification Office.

Note: Information concerning the existence of approved alternative methods of compliance with this airworthiness directive, if any, may be obtained from the Engine Certification Office.

(g) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the aircraft to a location where the requirements of this AD can be accomplished.

Issued in Burlington, Massachusetts, on April 1, 1996.

Jay J. Pardee,

Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. 96-11168 Filed 5-3-96; 8:45 am]

BILLING CODE 4910-13-U