

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. App. 1354(a), 1421 and 1423; 49 U.S.C. 106(g); and 14 CFR 11.89.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

95-12-07 Airbus Industrie: Amendment 39-9257. Docket 94-NM-142-AD.

Applicability: Model A340-211 and -311 airplanes on which Airbus Modification 42247 has not been installed (reference Airbus Service Bulletin A340-27-4013, dated October 27, 1993), certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must use the authority provided in paragraph (c) to request approval from the FAA. This approval may address either no action, if the current configuration eliminates the unsafe condition; or different actions necessary to address the unsafe condition described in this AD. Such a request should include an assessment of the effect of the changed configuration on the unsafe condition addressed by this AD. In no case does the presence of any modification, alteration, or repair remove any airplane from the applicability of this AD.

Compliance: Required as indicated, unless accomplished previously.

To prevent loss of hydraulic fluid, which may lead to the loss of the corresponding hydraulic system and its associated functions, and reduced controllability of the airplane, accomplish the following:

(a) Within 3 months after the effective date of this AD, replace the left- and right-hand inboard and outboard aileron servo controls associated with the green hydraulic system with new units that contain an improved seal installation, in accordance with Airbus Service Bulletin A340-27-4013, dated October 27, 1993.

(b) Within 6 months after the effective date of this AD, replace the left- and right-hand inboard and outboard aileron servo controls associated with the yellow and blue hydraulic systems with new units that contain an improved seal installation, in accordance with Airbus Service Bulletin A340-27-4013, dated October 27, 1993.

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Standardization Branch, ANM-113, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then

send it to the Manager, Standardization Branch, ANM-113.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Standardization Branch, ANM-113.

(d) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(e) The replacements shall be done in accordance with Airbus Service Bulletin A340-27-4013, dated October 27, 1993. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street NW., suite 700, Washington, DC.

(f) This amendment becomes effective on June 28, 1995.

Issued in Renton, Washington, on June 1, 1995.

Darrell M. Pederson,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 95-13889 Filed 6-12-95; 8:45 am]

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14 CFR Part 39

[Docket No. 95-NM-45-AD; Amendment 39-9259; AD 95-12-09]

Airworthiness Directives; Airbus Model A300 B4-1C, B4-2C, B4-203 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Airbus Model A300 series airplanes. This action requires repetitive inspections to detect cracking in the hinge fittings of the nose landing gear (NLG) aft doors, and replacement of cracked fittings. This amendment is prompted by several reports of loss of an NLG aft door during landing, due to failure of the door's hinge fittings. The actions specified in this AD are intended to prevent the loss of an NLG aft door due to the failure of the hinge fittings; such loss of a door can result in damage to the surrounding aircraft structure or injury to persons on the ground.

DATES: Effective on June 28, 1995.

The incorporation by reference of certain publications listed in the

regulations is approved by the Director of the Federal Register as of June 28, 1995.

Comments for inclusion in the Rules Docket must be received on or before August 14, 1995.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-103, Attention: Rules Docket No. 95-NM-45-AD, 1601 Lind Avenue SW., Renton, Washington 98055-4056.

The service information referenced in this AD may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Stephen Slotte, Aerospace Engineer, Standardization Branch, ANM-113, FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, Washington 98055-4056; telephone (206) 227-2797; fax (206) 227-1320.

SUPPLEMENTARY INFORMATION: The Direction Générale de l'Aviation Civile (DGAC), which is the airworthiness authority for France, recently notified the FAA that an unsafe condition may exist on certain Airbus Model A300 B4-1C, B4-2C, and B4-203 series airplanes. The DGAC advises that there have been four incidents in which an aft (secondary) door of the nose landing gear (NLG) on in-service airplanes was lost during landing. The doors separated from the airplanes due to rupture of the doors' forward hinge fitting. The cause of the fitting failures has been attributed to fatigue cracking. Such cracking, if not detected and corrected in a timely manner, can lead to separation of the NLG aft door from the airplane. Loss of a door can result in damage to the surrounding aircraft structure or injury to persons on the ground.

Airbus Industrie has issued Service Bulletin A300-52-0161, dated October 3, 1994, which describes procedures for performing repetitive eddy current inspections of the NLG aft door hinge fittings. This service bulletin also describes procedures for replacing cracked fittings. The DGAC approved this service bulletin as mandatory in order to assure the continued airworthiness of these airplanes in France.

This airplane model is manufactured in France and is type certificated for operation in the United States under the provisions of section 21.29 of the

Federal Aviation Regulations (14 CFR 21.19) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, the DGAC has kept the FAA informed of the situation described above. The FAA has examined the findings of the DGAC, reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same type design, this AD is being issued to prevent the loss of an NLG aft door due to the failure of the door's hinge fittings. This AD requires repetitive eddy current inspections to detect cracking of the hinge fittings, and replacement of any hinge fitting found to be cracked. Inspections continue after replacement of a hinge fitting is accomplished. The actions are required to be accomplished in accordance with the service bulletin described previously.

As a result of recent communications with the Air Transport Association (ATA) of America, the FAA has learned that, in general, some operators may misunderstand the legal effect of AD's on airplanes that are identified in the applicability provision of the AD, but that have been altered or repaired in the area addressed by the AD. The FAA points out that all airplanes identified in the applicability provision of an AD are legally subject to the AD. If an airplane has been altered or repaired in the affected area in such a way as to affect compliance with the AD, the owner or operator is required to obtain FAA approval for an alternative method of compliance with the AD, in accordance with the paragraph of each AD that provides for such approvals. A note has been included in this AD to clarify this long-standing requirement.

There currently are no affected Model A300 B4-C, B4-2C, or B4-203 series airplanes on the U.S. Register. All airplanes included in the applicability of this rule currently are operated by non-U.S. operators under foreign registry; therefore, they are not directly affected by this AD action. However, the FAA considers that this rule is necessary to ensure that the unsafe condition is addressed in the event that any of these subject airplanes are imported and placed on the U.S. Register in the future.

Should an affected airplane be imported and placed on the U.S. Register in the future, it would require approximately 3 work hours to accomplish the required actions, at an average labor charge of \$60 per work

hour. Based on these figures, the total cost impact of this AD would be \$180 per airplane per inspection cycle.

Since this AD action does not affect any airplane that is currently on the U.S. register, it has no adverse economic impact and imposes no additional burden on any person. Therefore, notice and public procedures hereon are unnecessary and the amendment may be made effective in less than 30 days after publication in the **Federal Register**.

Comments Invited

Although this action is in the form of a final rule and was not preceded by notice and opportunity for public comment, comments are invited on this rule. Interested persons are invited to comment on this rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified under the caption **ADDRESSES**. All communications received on or before the closing date for comments will be considered, and this rule may be amended in light of the comments received. Factual information that supports the commenter's ideas and suggestions is extremely helpful in evaluating the effectiveness of the AD action and determining whether additional rulemaking action would be needed.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the rule that might suggest a need to modify the rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report that summarizes each FAA-public contact concerned with the substance of this AD will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this rule must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 95-NM-45-AD." The postcard will be date stamped and returned to the commenter.

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this final rule does not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. App. 1354(a), 1421 and 1423; 49 U.S.C. 106(g); and 14 CFR 11.89.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

95-12-09 Airbus Industrie: Amendment 39-9259. Docket 95-NM-45-AD.

Applicability: Model A300 B4-1C, B4-2C and B4-203 series airplanes; having serial numbers 002 through 019, inclusive; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must use the authority provided in paragraph (d) to request approval from the FAA. This approval may address either no action, if the current configuration eliminates the unsafe condition; or different actions necessary to address the unsafe condition described in this AD. Such a request should include an assessment of the effect of the changed configuration on the unsafe condition addressed by this AD. In no case does the presence of any modification, alteration, or repair remove any airplane from the applicability of this AD.

Compliance: Required as indicated, unless accomplished previously.

To prevent the loss of a nose landing gear (NLG) aft door due to the failure of the door's hinge fittings, which can result in damage to the surrounding aircraft structure or injury to persons on the ground, accomplish the following:

(a) Prior to the accumulation of 8,000 total flight cycles, or within 500 flight cycles after the effective date of this AD, whichever occurs later, perform an eddy current inspection to detect cracks in the hinge fitting of the NLG left- and right-hand aft doors, in accordance with Airbus Service Bulletin A300-52-0161, dated October 3, 1994.

(b) If no crack(s) is found during the inspection required by paragraph (a) of this AD, repeat the eddy current inspection thereafter at intervals not to exceed 700 flight cycles.

(c) If any crack is found during the inspection required by paragraph (a) of this AD, prior to further flight, replace the hinge fittings in accordance with Airbus Service Bulletin A300-52-0161, dated October 3, 1994. Within 8,000 flight cycles after this replacement, and thereafter at intervals not to exceed 700 flight cycles, perform an eddy current inspection to detect cracking of the hinge fittings, in accordance with the service bulletin.

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Standardization Branch, ANM-113, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Standardization Branch, ANM-113.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Standardization Branch.

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(f) The inspection and replacement actions shall be done in accordance with Airbus Service Bulletin A300-52-0161, dated October 3, 1994. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street NW., suite 700, Washington, DC.

(g) This amendment becomes effective on June 28, 1995.

Issued in Renton, Washington, on June 1, 1995.

Darrell M. Pederson,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 95-13890 Filed 6-12-95; 8:45 am]

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14 CFR Part 39

[Docket No. 93-ANE-44; Amendment 39-9271; AD 94-01-03 R2]

Airworthiness Directives; Teledyne Continental Motors (Formerly Bendix) S-20, S-200, S-600, and S-1200 Series Magnetos

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments.

SUMMARY: This amendment revises an existing airworthiness directive (AD), applicable to Teledyne Continental Motors (TCM) (formerly Bendix) S-20, S-200, S-600, and S-1200 series magnetos, that currently requires replacing Bendix ignition coils and rotating magnets, regardless of total time in service (TIS), with improved TCM ignition coils, rotating magnets and marking magnetos to indicate compliance, except for the S-1200 series magnetos on which the AD requires replacing only the ignition coils as that series magneto already incorporates rotating magnets with the improved TCM design. This amendment removes several notes after the applicability paragraph and inserts these as paragraphs into the applicability itself and into the compliance section to clarify that these actions are mandatory. In addition, this amendment allows installation of replacement serviceable Parts Manufacturer Approval (PMA) parts in addition to TCM parts. Also, this amendment clarifies that Bendix magnetos replaced with Slick magnetos satisfy the requirements of the AD, and that operators must perform the requirements of the AD on magnetos with Bendix magneto data plates that have been replaced with an overhaul facility's data plate. This amendment is prompted by comments that request clarification of the compliance notes and by the request to install replacement serviceable PMA parts. The actions specified by this AD are intended to prevent magneto failure and subsequent engine failure.

DATES: Effective on June 28, 1995.

The incorporation by reference of certain publications listed in the regulations was previously approved by

the Director of the Federal Register as of September 6, 1994 (59 FR 43029, August 22, 1994).

Comments for inclusion in the Rules Docket must be received on or before August 14, 1995.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), New England Region, Office of the Assistant Chief Counsel, Attention: Rules Docket No. 93-ANE-44, 12 New England Executive Park, Burlington, MA 01803-5299.

The service information referenced in this AD may be obtained from Teledyne Continental Motors, P.O. Box 90, Mobile, AL 36601; telephone (205) 438-3411. This information may be examined at the FAA, New England Region, Office of the Assistant Chief Counsel, 12 New England Executive Park, Burlington, MA; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC. **FOR FURTHER INFORMATION CONTACT:** Jerry Robinette, Aerospace Engineer, Atlanta Aircraft Certification Office, FAA, Small Airplane Directorate, Campus Building, 1701 Columbia Ave., Suite S-160, College Park, GA 30337-2748; telephone (404) 305-7371; fax (404) 305-7348.

SUPPLEMENTARY INFORMATION: On December 29, 1993, the Federal Aviation Administration (FAA) issued airworthiness directive (AD) 94-01-03, Amendment 39-8785 (59 FR 4555, February 1, 1994), to require replacing certain Bendix ignition coils and rotating magnets, regardless of time in service (TIS), with improved serviceable Teledyne Continental Motors (TCM) ignition coils and rotating magnets at either the next 100-hour inspection, the next annual inspection, the next progressive inspection, or the next 100 hours TIS after the effective date of the AD, whichever occurs first. For S-1200 series magnetos, the AD requires replacing only the ignition coils as the rotating magnets on that series magneto already incorporates the improved TCM design. Additionally, the AD requires re-marking magnetos to indicate compliance. That action was prompted by reports of accidents caused by failures of magnetos incorporating older Bendix components that had not been replaced in accordance with superseded AD 73-07-04, Amendment 39-1731 (38 FR 27600, October 5, 1973). That condition, if not corrected, could result in magneto failure and subsequent engine failure.

On August 11, 1994, the FAA issued AD 94-17-11, Amendment 39-9006 (59 FR 43029, August 22, 1994), that revises AD 94-01-03, and notes that an error in