

accumulation of 21,600 flight cycles, perform an HFEC inspection to detect cracks at the aft spar web of the wings, in accordance with Airbus Service Bulletin A300-57-6059, dated August 12, 1994. Repeat the inspection thereafter at intervals not to exceed 5,700 flight cycles.

(e) If any crack is detected during any inspection required by this AD, prior to further flight, repair the crack in accordance with Airbus Service Bulletin A300-57-0213, dated August 12, 1994, or Airbus Service Bulletin A300-57-6059, dated August 12, 1994, as applicable; or in accordance with a method approved by the Manager, Standardization Branch, ANM-113, FAA, Transport Airplane Directorate.

(f) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Standardization Branch, ANM-113, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Standardization Branch, ANM-113.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Standardization Branch, ANM-113.

(g) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Issued in Renton, Washington, on April 28, 1995.

**James V. Devany,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 95-10988 Filed 5-3-95; 8:45 am]

BILLING CODE 4910-13-U

## 14 CFR Part 39

[Docket No. 95-NM-31-AD]

### Airworthiness Directives; Beech Model 400 and 400A Airplanes

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain Beech Model 400 and 400A airplanes. This proposal would require modification of the autopilot and rudder boost interlock. This proposal is prompted by a report indicating that the rudder boost system installed on these airplanes does not operate correctly during deployment of a thrust reverser. The actions specified by the proposed AD are intended to prevent incorrect

operation of the rudder boost system during deployment of a thrust reverser and to prevent the autopilot from exceeding certain bank angle limits; these conditions could result in reduced controllability of the airplane.

**DATES:** Comments must be received by June 13, 1995.

**ADDRESSES:** Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-103, Attention: Rules Docket No. 95-NM-31-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9:00 a.m. and 3:00 p.m., Monday through Friday, except Federal holidays.

The service information referenced in the proposed rule may be obtained from Beech Aircraft Corporation, Commercial Service Department, P.O. Box 85, Wichita, Kansas 67201-0085. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Small Airplane Directorate, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Mid-Continent Airport, Wichita, Kansas.

**FOR FURTHER INFORMATION CONTACT:** Dale Vassalli, Aerospace Engineer, Systems and Equipment Branch, ACE-130W, FAA, Wichita Aircraft Certification Office, Small Airplane Directorate, 1801 Airport Road, Room 100, Mid-Continent Airport, Wichita, Kansas 67209; telephone (316) 946-4132; fax (316) 946-4407.

#### SUPPLEMENTARY INFORMATION:

##### Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this notice may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this

proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 95-NM-31-AD." The postcard will be date stamped and returned to the commenter.

#### Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-103, Attention: Rules Docket No. 95-NM-31-AD, 1601 Lind Avenue SW., Renton, Washington 98055-4056.

#### Discussion

The FAA received a report from the airplane manufacturer indicating that, during ground operation, the rudder boost system installed on Beech Model 400 and 400A airplanes is disabled only when the left thrust reverser is deployed. Operation of either the right or left thrust reverser during ground operation should disable the rudder boost system. Additionally, during flight, the rudder boost system on these airplanes is disabled when inadvertent deployment of the left thrust reverser occurs. However, inadvertent deployment of a thrust reverser should not disable the rudder boost system.

The FAA also discovered that, when landing the airplane with a failed left engine, use of the right thrust reverser will result in a rudder boost in the wrong direction. (When landing with a failed right engine, use of the left thrust reverser will disable the rudder boost system, as it should.) Further, inadvertent deployment of the left thrust reverser will result in disengagement of the rudder boost system. Should this condition occur during takeoff, rudder forces could exceed the limits specified in the Federal Aviation Regulations (FAR).

These conditions, if not corrected, could result in reduced controllability of the airplane.

The FAA has reviewed and approved Beechcraft Service Bulletin No. 2533, dated October 1994, which describes procedures for modification of the autopilot and rudder boost interlock. The modification entails installing an autopilot and rudder boost improvement kit. Installation of the kit will disable the rudder boost feature during operation of the thrust reverser with only one engine operating in order to alleviate control input requirements. In addition, the service bulletin describes removal of a placard if one

was installed in accordance with Beechcraft Service Bulletin No. 2502, dated May 1993.

Since an unsafe condition has been identified that is likely to exist or develop on other products of this same type design, the proposed AD would require modification of the autopilot and rudder boost interlock. The actions would be required to be accomplished in accordance with the service bulletin described previously.

As a result of recent communications with the Air Transport Association (ATA) of America, the FAA has learned that, in general, some operators may misunderstand the legal effect of AD's on airplanes that are identified in the applicability provision of the AD, but that have been altered or repaired in the area addressed by the AD. The FAA points out that all airplanes identified in the applicability provision of an AD are legally subject to the AD. If an airplane has been altered or repaired in the affected area in such a way as to affect compliance with the AD, the owner or operator is required to obtain FAA approval for an alternative method of compliance with the AD, in accordance with the paragraph of each AD that provides for such approvals. A note has been included in this notice to clarify this long-standing requirement.

There are approximately 92 Model 400 and 400A airplanes of the affected design in the worldwide fleet. The FAA estimates that 69 airplanes of U.S. registry would be affected by this proposed AD, that it would take approximately 24 work hours per airplane to accomplish the proposed actions, and that the average labor rate is \$60 per work hour. Required parts would be provided by the manufacturer at no cost to operators. Based on these figures, the total cost impact of the proposed AD on U.S. operators is estimated to be \$99,360, or \$1,440 per airplane.

The total cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

The regulations proposed herein would not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this proposal would not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption

**ADDRESSES.**

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

#### The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. App. 1354(a), 1421 and 1423; 49 U.S.C. 106(g); and 14 CFR 11.89.

##### § 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

**Beech Aircraft Corporation:** Docket 95–NM–31–AD.

**Applicability:** Model 400 airplanes, serial RJ–61; and Model 400A airplanes, serials RK–1 through RK–77 inclusive, and RK–79 through RK–92 inclusive; certificated in any category.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must use the authority provided in paragraph (b) to request approval from the FAA. This approval may address either no action, if the current configuration eliminates the unsafe condition; or different actions necessary to address the unsafe condition described in this AD. Such a request should include an assessment of the effect of the changed configuration on the unsafe condition addressed by this AD. In no case does the presence of any modification, alteration, or repair remove any airplane from the applicability of this AD.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent reduced controllability of the airplane, accomplish the following:

(a) At the next scheduled inspection, but no later than 200 hours time-in-service after the effective date of this AD, install an autopilot and rudder boost improvement kit in accordance with Beechcraft Service Bulletin No. 2533, dated October 1994.

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Wichita Aircraft Certification Office (ACO), FAA, Small Airplane Directorate.

Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Wichita ACO.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Wichita ACO.

(c) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished. Issued in Renton, Washington, on April 28, 1995.

**James V. Devany,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 95–10989 Filed 5–3–95; 8:45 am]

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#### Coast Guard

#### 33 CFR Part 117

[CGD01–95–009]

RIN 2115–AE47

#### Drawbridge Operation Regulations; Connecticut River, CT

**AGENCY:** Coast Guard, DOT.

**ACTION:** Notice of proposed rulemaking.

**SUMMARY:** The Coast Guard is proposing a change to the regulations for the Route 82 Bridge at mile 16.8 over the Connecticut River, between East Haddam and Haddam, Connecticut. This proposal would provide openings for recreational vessels only on the hour and half-hour from 15 May through 31 October, between 9 a.m. and 9 p.m. Commercial vessels would continue to be granted bridge openings at all times. This action should ease traffic delays and still meet the reasonable needs of navigation.

This proposal would also require bridge owners to install clearance gauges at the AMTRAK Old Saybrook-Old Lyme Bridge, the CONRAIL Middletown-Portland Bridge and the Route 82 Bridge to assist mariners in