percent for three-engine airplanes, and 1.7 percent for four-engine airplanes with—

(i) The critical engine inoperative and its propeller in the minimum drag position;
(ii) The remaining engine(s) at not more than maximum continuous power;
(iii) The landing gear retracted;
(iv) The wing flaps retracted; and
(v) A climb speed not less than 1.2 $V_{S_1}$.

(4) **Discontinued approach.** The steady gradient of climb at an altitude of 400 feet above the landing surface must be not less than 2.1 percent for two-engine airplanes, 2.4 percent for three-engine airplanes, and 2.7 percent for four-engine airplanes, with—

(i) The critical engine inoperative and its propeller in the minimum drag position;
(ii) The remaining engine(s) at takeoff power;
(iii) Landing gear retracted;
(iv) Wing flaps in the approach position(s) in which $V_{S_1}$ for these position(s) does not exceed 110 percent of the $V_{S_1}$ for the related all-engines-operated landing position(s); and
(v) A climb speed established in connection with normal landing procedures but not exceeding 1.5 $V_{S_1}$.

[Doc. No. 27807, 61 FR 5187, Feb. 9, 1996]  


The maximum horizontal distance traveled in still air, in nautical miles, per 1,000 feet of altitude lost in a glide, and the speed necessary to achieve this must be determined with the engine inoperative, its propeller in the minimum drag position, and landing gear and wing flaps in the most favorable available position.

[Doc. No. 27807, 61 FR 5187, Feb. 9, 1996]  

§ 23.73 Reference landing approach speed.  

(a) For normal, utility, and acrobatic category reciprocating engine-powered airplanes of 6,000 pounds or less maximum weight, the reference landing approach speed, $V_{REF}$, may not be less than the greater of $V_{MC}$, determined in §23.149(b) with the wing flaps in the most extended takeoff position, and 1.3 $V_{S_1}$.

(b) For normal, utility, and acrobatic category turbine powered airplanes of 6,000 pounds or less maximum weight, turboprops of more than 6,000 pounds maximum weight, and reciprocating engine-powered airplanes of more than 6,000 pounds maximum weight, the reference landing approach speed, $V_{REF}$, may not be less than the greater of $V_{MC}$, determined in §23.149(c), and 1.3 $V_{S_1}$.

(c) For normal, utility, and acrobatic category jets of more than 6,000 pounds maximum weight and commuter category airplanes, the reference landing approach speed, $V_{REF}$, may not be less than the greater of 1.05 $V_{MC}$, determined in §23.149(c), and 1.3 $V_{S_1}$.


§ 23.75 Landing distance.  

The horizontal distance necessary to land and come to a complete stop from