§ 27.361 Engine torque.  
(a) For turbine engines, the limit torque may not be less than the highest of—  
(1) The mean torque for maximum continuous power multiplied by 1.25;  
(2) The torque required by § 27.923;  
(3) The torque required by § 27.927; or  
(4) The torque imposed by sudden engine stoppage due to malfunction or structural failure (such as compressor jamming).  

(b) For reciprocating engines, the limit torque may not be less than the mean torque for maximum continuous power multiplied by—  
(1) 1.33, for engines with five or more cylinders; and  
(2) Two, three, and four, for engines with four, three, and two cylinders, respectively.  


§ 27.391 General.  
Each auxiliary rotor, each fixed or movable stabilizing or control surface, and each system operating any flight control must meet the requirements of §§ 27.395, 27.397, 27.411, and 27.427.  


§ 27.397 Limit pilot forces and torques.  
(a) Except as provided in paragraph (b) of this section, the limit pilot forces are as follows:  
(1) For foot controls, 130 pounds.  
(2) For stick controls, 100 pounds fore and aft, and 67 pounds laterally.  
(b) For flap, tab, stabilizer, rotor brake, and landing gear operating controls, the forces apply (R=radius in inches):  
(1) Crank, wheel, and lever controls, [1+R]/3 × 50 pounds, but not less than 50 pounds nor more than 100 pounds for hand operated controls or 120 pounds for foot operated controls, applied at
$27.399$ Dual control system.

Each dual primary flight control system must be designed to withstand the loads that result when pilot forces of $0.75$ times those obtained under §27.395 are applied—

(a) In opposition; and

(b) In the same direction.

$27.411$ Ground clearance: tail rotor guard.

(a) It must be impossible for the tail rotor to contact the landing surface during a normal landing.

(b) If a tail rotor guard is required to show compliance with paragraph (a) of this section—

(1) Suitable design loads must be established for the guard; and

(2) The guard and its supporting structure must be designed to withstand those loads.

$27.427$ Unsymmetrical loads.

(a) Horizontal tail surfaces and their supporting structure must be designed for unsymmetrical loads arising from yawing and rotor wake effects in combination with the prescribed flight conditions.

(b) To meet the design criteria of paragraph (a) of this section, in the absence of more rational data, both of the following must be met:

(1) One hundred percent of the maximum loading from the symmetrical flight conditions acts on the surface on one side of the plane of symmetry, and no loading acts on the other side.

(2) Fifty percent of the maximum loading from the symmetrical flight conditions acts on the surface on each side of the plane of symmetry but in opposite directions.

(c) For empennage arrangements where the horizontal tail surfaces are supported by the vertical tail surfaces, the vertical tail surfaces and supporting structure must be designed for the combined vertical and horizontal surface loads resulting from each prescribed flight condition, considered separately. The flight conditions must be selected so the maximum design loads are obtained on each surface. In the absence of more rational data, the unsymmetrical horizontal tail surface loading distributions described in this section must be assumed.

$27.471$ General.

(a) Loads and equilibrium. For limit ground loads—

(1) The limit ground loads obtained in the landing conditions in this part must be considered to be external loads that would occur in the rotorcraft structure if it were acting as a rigid body; and

(2) In each specified landing condition, the external loads must be placed in equilibrium with linear and angular inertia loads in a rational or conservative manner.

(b) Critical centers of gravity. The critical centers of gravity within the range for which certification is requested must be selected so that the maximum design loads are obtained in each landing gear element.

$27.473$ Ground loading conditions and assumptions.

(a) For specified landing conditions, a design maximum weight must be used that is not less than the maximum weight. A rotor lift may be assumed to act through the center of gravity throughout the landing impact. This lift may not exceed two-thirds of the design maximum weight.

(b) Unless otherwise prescribed, for each specified landing condition, the rotorcraft must be designed for a limit load factor of not less than the limit inertia load factor substantiated under §27.725.

$27.475$ Tires and shock absorbers.

Unless otherwise prescribed, for each specified landing condition, the tires must be assumed to be in their static position and the shock absorbers to be in their most critical position.