§ 23.365 Pressurized cabin loads.
For each pressurized compartment, the following apply:
(a) The airplane structure must be strong enough to withstand the flight loads combined with pressure differential loads from zero up to the maximum relief valve setting.
(b) The external pressure distribution in flight, and any stress concentrations, must be accounted for.
(c) If landings may be made with the cabin pressurized, landing loads must be combined with pressure differential loads from zero up to the maximum allowed during landing.
(d) The airplane structure must be strong enough to withstand the pressure differential loads corresponding to the maximum relief valve setting multiplied by a factor of 1.33, omitting other loads.
(e) If a pressurized cabin has two or more compartments separated by bulkheads or a floor, the primary structure must be designed for the effects of sudden release of pressure in any compartment with external doors or windows. This condition must be investigated for the effects of failure of the largest opening in the compartment. The effects of intercompartmental venting may be considered.

§ 23.367 Unsymmetrical loads due to engine failure.
(a) Turbopropeller airplanes must be designed for the unsymmetrical loads resulting from the failure of the critical engine including the following conditions in combination with a single malfunction of the propeller drag limiting system, considering the probable pilot corrective action on the flight controls:
(1) At speeds between \( V_{MC} \) and \( V_P \), the loads resulting from power failure because of fuel flow interruption are considered to be limit loads.
(2) At speeds between \( V_{MC} \) and \( V_C \), the loads resulting from the disconnection of the engine compressor from the turbine or from loss of the turbine blades are considered to be ultimate loads.
(b) The side load prescribed in paragraph (a) of this section may be assumed to be independent of other flight conditions.

§ 23.369 Rear lift truss.
(a) If a rear lift truss is used, it must be designed to withstand conditions of reversed airflow at a design speed of—
\[ V = 8.7 \sqrt{\frac{W}{S}} + 8.7 \text{ knots}, \]
where \( W/S \) = wing loading at design maximum takeoff weight.
(b) Either aerodynamic data for the particular wing section used, or a value of \( C_L \) equalling 0.8 with a chordwise distribution that is triangular between a peak at the trailing edge and zero at the leading edge, must be used.

§ 23.371 Gyroscopic and aerodynamic loads.
(a) Each engine mount and its supporting structure must be designed for the gyroscopic, inertial, and aerodynamic loads that result, with the engine(s) and propeller(s), if applicable, at maximum continuous r.p.m., under either:
(1) The conditions prescribed in § 23.351 and § 23.423; or