§ 23.1095 Carburetor deicing fluid flow rate.
(a) If a carburetor deicing fluid system is used, it must be able to simultaneously supply each engine with a rate of fluid flow, expressed in pounds per hour, of not less than 2.5 times the square root of the maximum continuous power of the engine.
(b) The fluid must be introduced into the air induction system—
   (1) Close to, and upstream of, the carburetor; and
   (2) So that it is equally distributed over the entire cross section of the induction system air passages.
§ 23.1097 Carburetor deicing fluid system capacity.
(a) The capacity of each carburetor deicing fluid system—
   (1) May not be less than the greater of—
      (i) That required to provide fluid at the rate specified in §23.1095 for a time equal to three percent of the maximum endurance of the airplane; or
      (ii) 20 minutes at that flow rate; and
   (2) Need not exceed that required for two hours of operation.
(b) If the available preheat exceeds 50 °F, but is less than 100 °F, the capacity of the system may be decreased in proportion to the heat rise available in excess of 50 °F.
§ 23.1099 Carburetor deicing fluid system detail design.
Each carburetor deicing fluid system must meet the applicable requirements for the design of a fuel system, except as specified in §§23.1095 and 23.1097.
§ 23.1101 Induction air preheater design.
Each exhaust-heated, induction air preheater must be designed and constructed to—
(a) Ensure ventilation of the preheater when the induction air preheater is not being used during engine operation;
(b) Allow inspection of the exhaust manifold parts that it surrounds; and
(c) Allow inspection of critical parts of the preheater itself.
§ 23.1103 Induction system ducts.
(a) Each induction system duct must have a drain to prevent the accumulation of fuel or moisture in the normal ground and flight attitudes. No drain may discharge where it will cause a fire hazard.
(b) Each duct connected to components between which relative motion could exist must have means for flexibility.
(c) Each flexible induction system duct must be capable of withstanding the effects of temperature extremes, fuel, oil, water, and solvents to which it is expected to be exposed in service and maintenance without hazardous deterioration or delamination.
(d) For reciprocating engine installations, each induction system duct must be—
   (1) Fireproof within the auxiliary power unit compartment;
   (2) Fireproof for a sufficient distance upstream of the auxiliary power unit compartment to prevent hot gas reverse flow from burning through the duct and entering any other compartment of the airplane in which a hazard would be created by the entry of the hot gases;
   (3) Constructed of materials suitable to the environmental conditions expected in service, except in those areas requiring fireproof or fire resistant materials; and
   (4) Constructed of materials that will not absorb or trap hazardous quantities of flammable fluids that could be ignited by a surge or reverse-flow condition.
§ 23.1121  General.

For powerplant and auxiliary power unit installations, the following apply—

(a) Each exhaust system must ensure safe disposal of exhaust gases without fire hazard or carbon monoxide contamination in any personnel compartment.

(b) Each exhaust system part with a surface hot enough to ignite flammable fluids or vapors must be located or shielded so that leakage from any system carrying flammable fluids or vapors will not result in a fire caused by impingement of the fluids or vapors on any part of the exhaust system including shields for the exhaust system.

(c) Each exhaust system must be separated by fireproof shields from adjacent flammable parts of the airplane that are outside of the engine and auxiliary power unit compartments.

(d) No exhaust gases may discharge dangerously near any fuel or oil system drain.

(e) No exhaust gases may be discharged where they will cause a glare following any probable failure of the turbocharger or its lubrication system.

§ 23.1111  Turbine engine bleed air system.

For turbine engine bleed air systems, the following apply:

(a) No hazard may result if duct rupture or failure occurs anywhere between the engine port and the airplane unit served by the bleed air.

(b) The effect on airplane and engine performance of using maximum bleed air must be established.

(c) Hazardous contamination of cabin air systems may not result from failures of the engine lubricating system.

§ 23.1111  Turbine engine bleed air system.

For turbine engine bleed air systems, the following apply:

(a) No hazard may result if duct rupture or failure occurs anywhere between the engine port and the airplane unit served by the bleed air.

(b) The effect on airplane and engine performance of using maximum bleed air must be established.

(c) Hazardous contamination of cabin air systems may not result from failures of the engine lubricating system.